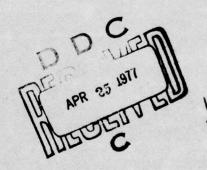


**TECHNICAL REPORT RD-77-13** 

AN INVESTIGATION OF STREAMWISE BODY-FIN GAPS AS A MEANS OF ALLEVIATING THE ADVERSE PLUME EFFECTS ON MISSILE LONGITUDINAL STABILITY

Aeroballistics Directorate
US Army Missile Research, Development and Engineering Laboratory
US Army Missile Command
Redstone Arsenal, Alabama 35809

January 1977



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U.S. ARMY MISSILE COMMAND

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Unclassified SECURITY CLASSIFICATION OF THIS PAGE (When Date Entered) READ INSTRUCTIONS REPORT DOCUMENTATION PAGE BEFORE COMPLETING FORM RECIPIENT'S CATALOG NUMBER 2. GOVT ACCESSION NO. 1. REPORT NUMBER Technical Report RD-77-13 YPE OF REPORT & PERIOD COVERED Technical Report, AN INVESTIGATION OF STREAMWISE BODY-FIN GAPS AS A MEANS OF ALLEVIATING THE ADVERSE PLUME PERFORMING ORG. RE EFFECTS ON MISSILE LONGITUDINAL STABILITY. DMS-AR-1038 CONTRACT OR GRANT NUMBER(+) Henderson DAAH01-76-C-0301 James H. PERFORMING ORGANIZATION NAME AND ADDRESS 10. PROGRAM ELEME AREA & WORK UN US Army Missile Command DA Project No. (1W362303A214 Attn: AMSMI-RDK AMC MS Code 632303.11.21400 Redstone Arsenal, Alabama 35809 CONTROLLING OFFICE NAME AND ADDRESS January 1977 US Army Missile Command
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MONITORING AGENCY NAME & ADDRESS(II different from Controlling Office) 87 15. SECURITY CLASS. (of this report) Unclassified 154. DECLASSIFICATION/DOWNGRADING 16. DISTRIBUTION STATEMENT (of this Report) Approved for public release; distribution unlimited. 17. DISTRIBUTION STATEMENT (of the abstract entered in Block 20, if different from Report) 18. SUPPLEMENTARY NOTES This report was prepared from data plotted by the Data Management Services, Chrysler Corporation Space Division. All data presented are in a data base N. 0 . maintained by Data Management Services and are available in tabular or magnetic tape form. 19. KEY WORDS (Continue on reverse side if necessary and identify by block number) Thrust Effects Jet Plume Longitudinal Stability Gap Effects Plume Effects Base Pressure Transonic wind tunnel tests were conducted on a body of revolution with various fin configurations to investigate jet plume effects on missile longitudinal stability. The sting-mounted model utilized a series of cold air normal jets located downstream of the base to simulate the jet plume. Fins of various planform geometry were tested in three longitudinal positions. Present report compares one fin planform with and without a 5 percent body diameter gap. Gap results in a small reduction of adverse plume effects for fins in aft DD 1 JAN 79 1473 EDITION OF ! NOV 65 IS OBSOLETE

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20. ABSTRACT (Concluded)

position. Data for this investigation were obtained from Calspan test CAL T17-123.



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PLOT SCHEDULE:

(A) C<sub>N</sub> VERSUS α
(B) C<sub>NF2</sub> VERSUS α
(C) C<sub>Nα</sub> VERSUS C<sub>RT</sub>
(D) C<sub>NαF2</sub> VERSUS C<sub>RT</sub>

#### INTRODUCTION

During the past few years, the Army Missile Command has been interested in the adverse effects of the propulsive jet plume on missile aerodynamics (see reference 1). Of particular importance are the effects on missile longitudinal stability. Wind tunnel investigations have been conducted to determine effect of thrust level on missile longitudinal stability and to assess the effectiveness of locating fins some distance forward of the base to alleviate adverse effects on stability (see references 2, 3, and 4). Results of these tests indicated that fin effectiveness (when affected by the plume) improved as the distance from the base was increased.

One of the fin planforms tested in reference 4 (F2) was also tested with a five percent of body diameter longitudinal gap at the fin-body juncture (F4). It was expected that the gap would reduce the two-dimensional flow effects at this juncture and alleviate the adverse plume effects on missile stability. Reference 4 presents results. In the present report, data for the F2 and F4 configurations are compared directly. Data at angles of attack higher than four degrees are excluded.

Past tests (for example, reference 6) have shown that gaps cause a loss in fin effectiveness. This loss is caused by a partial cancellation of the fin-body interference lift. The purpose of the test of reference 6 was to investigate the feasibility of using gaps to reduce the large increase in stability margin of rockets which occurs at transonic speeds. The reduction in transonic stability is one of the means of reducing rocket

wind sensitivity during boost. Some typical results from reference 6 are compared with present results.

#### APPARATUS AND TESTS

These tests were described in detail in references 4 and 5. Briefly the sting-mounted model, which had a 6-component strain gage balance, was 52 inches in length and 5 inches in diameter. It had a 3-caliber tangent ogive nose and a cylindrical body. The model was tested in combination with cruciform rectangular fins; each fin panel had a 3-component strain gage balance.

Plume simulator consisted of 24 normal sonic jets arranged circumferentially in two rows with a common air chamber. The simulator was located 2.5 inches aft of the base. The combined exit area of the 24 jets represented six percent of the model base (reference area).

Fin nomenclature is as follows:

F2 (BF2)--fins with 3-inch chord and 2.5-inch semi-span.

F4 (BF4)--fins with 3-inch chord, 2.5-inch semi-span, and a .25-inch gap between the fuselage and the root chord.

Fin positions are as follows:

Aft--fin trailing edge flush with model base (M.S. 52.000, 0 cal.) --fin hinge line at M.S. 49.750

Mid--fin trailing edge 3.75 inches from base (M.S. 48.250, 0.75 cal.)
--fin hinge line at M.S. 46.000

Fwd--fin trailing edge 7.5 inches from base (M.S. 44.500, 1.5 cal.) --fin hinge line at M.S. 42.250

Model installation is shown in figure 3. A description of the geometry for fins F2 and F4 is shown in figure 4.

#### TEST CONDITIONS

The test was conducted in the Calspan Corporation 8-foot Transonic Wind Tunnel. A detailed description of this facility is presented in reference 5. Test Mach numbers ranged from 0.4 to 1.25. The amount of air in the tunnel circuit was equivalent to a wind-off static pressure of 0.5 atmosphere. Plume effect, or value of  $C_{RT}$ , was varied by changing simulator chamber pressure  $P_{C}$ . Normally, angle of attack is varied during a run and configuration,  $C_{RT}$ , and tunnel operating conditions are kept constant. Air was supplied to the plume simulator at flow rates up to 15 pounds per second and at chamber pressures to 600 psi. Angles of attack were varied from -4 to 12 degrees.

Model aerodynamic data were resolved in the body axis system with the origin located at model station 26.5 on the model centerline. Aerodynamic data from the tail fin balances were resolved in individual axis systems; origins were located at the intersection of the tail fin lines and the balance centerlines.

#### PLUME SIMULATION

Base pressure ratio  $P_b/P_\infty$  is a good measure of plume effects on missile aerodynamics (see reference 1). One of the parameters convenient to use with base pressure is thrust coefficient  $C_T$ , where  $C_T$  is axial thrust non-dimensionalized by dynamic pressure and body cross sectional area S. For the normal jet simulator, a similar parameter is radial thrust coef-

ficient CRT, where

$$C_{RT} = \frac{\text{radial thrust}}{qS}$$

For an axial jet, base pressure appears to be primarily influenced by the portion of the jet plume in the vicinity where the jet boundary interacts with the freestream flow. Where C<sub>T</sub> can be considered to represent the axial component of the effective jet, it can be assumed that C<sub>RT</sub> represents the normal component.

For the plume size of interest in the current investigation, a value of CT several times the value of CRT is required for matching base pressures. The exact CT/CRT ratio will depend on a comparison of flight base pressures with base pressure value for the normal jet simulator.

#### RESULTS AND DISCUSSION

Plotted data consist of comparisons between BF2 and BF4 total normal forces, C<sub>N</sub>, and fin no. 2 panel normal forces, C<sub>NF2</sub>, for various levels of simulated thrust. Matching runs for BF2 and BF4 were made from Mach 0.7 to 1.2 for zero roll position and from Mach 0.9 to 1.2 for a roll of 45 degrees. Panel normal force is presented for fin no. 2 only.

Inspection of the plots of  $C_N$  and  $C_{NF2}$  versus angle of attack generally shows a small but definite effect of the gap. A better overall view of gap effects may be obtained from plots of  $C_{N_{\alpha}}$  and  $C_{N_{\alpha}F2}$  versus  $C_{RT}$  with no plume effects ( $C_{RT} \leq .1$ ). As expected, a small decrement is apparent.

Values of  $C_N$  for fins in the aft position are compared with data of reference 6 in figure 5. In this comparison, body alone  $C_N$  was subtracted from the present data since afterbody forces only were measured in reference 6. Model geometry of reference 6 is shown in figures 6 and 7. Although geometry and gap size differ between the two tests, the gap effects for small gaps is about the same.

With plume effects, there is a small but definite improvement in  $C_{N_{\alpha}}$  and  $C_{N_{\alpha}F2}$  with fins in the aft position. With fins in the mid position, most of the decrement in  $C_{N_{\alpha}}$  without plume effects is canceled when there are plume effects. There appears to be no change with plume effects due to the gap with fins in the forward position.

For fins in the aft position, it is possible for a larger gap size to result in a significant reduction in adverse plume effects on fin effectiveness. In cases where missile design considerations force the use of aft fins, it may be profitable to investigate gap and fin geometries.

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## NOMENCLATURE

SYMBOL	MNENOMIC	DEFINITION
RN/L	RN/L	unit Reynolds number; per ft
٧		velocity; ft/sec
α	ALPHA	angle of attack, degrees
β	BETA	angle of sideslip, degrees
ψ	PSI	angle of yaw, degrees
ф	PHI	angle of roll, degrees
ρ		mass density; slugs/ft <sup>3</sup>

# Reference & C.G. Definitions

Ab		base area; m <sup>2</sup> , in <sup>2</sup>
b,bref	BREF	wing span or reference span; m, in
c.g.		center of gravity
ℓREF, c	LREF	reference length or wing mean aerodynamic chord; m, in
P <sub>C</sub>		simulator chamber pressure
S,S <sub>ref</sub>	SREF	reference area based on body diameter, in <sup>2</sup>
	MRP	moment reference point
	XMRP	moment reference point on X axis
	YMRP	moment reference point on Y axis
	ZMRP	moment reference point on Z axis

## NOMENCLATURE (Continued)

SYMBOL	MNEMONIC	DEFINITION
	Bo	ody Axis System
CN	CN	normal force coefficient; FN/qS
CA	CA	axial force coefficient; FA/qS
$c_{Ab}$	CAB	base axial force coefficient; $(-1) \cdot [(p_b - p_\infty)/q] (A_b/S)$
CAu	CAU	uncorrected axial force coefficient
Cm	CLM	pitching moment coefficient; MY/qS &ref
Су	CY	side force coefficient; Fy/qS
Cn	CYN	yawing moment coefficient; Mz/qS bref
Ce	CBL	rolling moment coefficient; M <sub>X</sub> /qS b <sub>ref</sub>
C <sub>NTX</sub>	CNTX	fin normal force coefficient for fin number X Fin Normal Force qS
C <sub>NFX</sub>	CNFX	fin number X corrected normal force coefficient,  CNF1 = CNT1  CNF2 = CNT2  CNF3 = (-1.0) CNT3  CNF4 = (-1.0) CNT4
СНМЕХ	CLMHX	fin hinge moment coefficient,  Fin Hinge Moment  qS lref
C <sub>BMFX</sub>	CLMRX	root bending moment coefficient,  Fin Root Bending Moment  qS lref

# NOMENCLATURE (Continued)

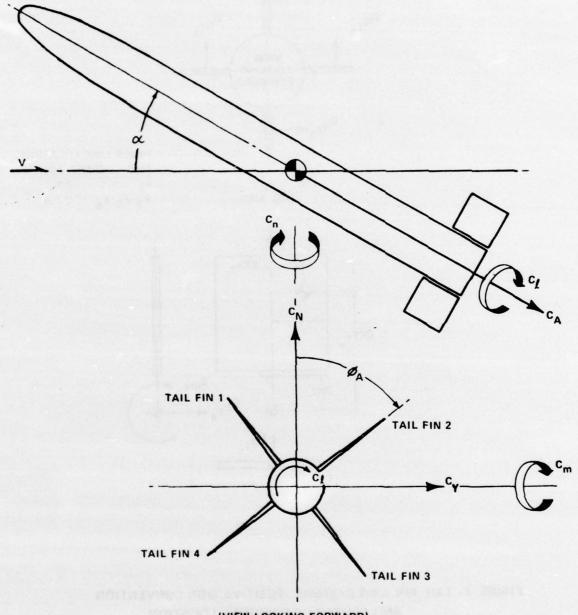
SYMBOL	MNENONIC	DEFINITION
	CPB1	base pressure coefficient at tap 1
	CPB2	base pressure coefficient at tap 2
	CPC	cavity pressure coefficient
	CPXT1	tail fin number 1 center of pressure location
	CPYT1	tail fin number 1 lateral center of pressure location
CT	СТ	thrust coefficient, axial thrust/qS
CRT	CRT	radial thrust coefficient, radial thrust/qS
$c_{N_{\alpha}}$	CNA	normal force coefficient derivative with respect to angle of attack
$c_{N_{\alpha FX}}$	CNAFX	fin normal force coefficient derivative with respect to angle of attack
$P_{b1}/P_{\infty}$	PB1/P	base pressure tap 1 ratio to freestream pressure
$P_{b_2}/P_{\infty}$	PB2/P	base pressure tap 2 ratio to freestream pressure
$\frac{(P_{b1}/P_{\infty}+P_{b2}/P_{\infty})}{2}$	PB-AVG	average of two base pressure tap ratios
Fin Position	FINPOS	fin position on body; defined on page 6
	PL	air supply pressure upstream of orifice, psi
	WA	air supply flow rate, lb/sec
$c_{m_{\alpha}}$		pitching moment coefficient derivative with respect to angle of attack

# NOMENCLATURE (Concluded)

SYMBOL	MNEMONIC	DEFINITION
a		speed of sound; ft/sec
Cp	CP	pressure coefficient; $(p_1 - p_{\infty})/q$
M smalphy	MACH	Mach number; V/a
p		pressure; psf
q	Q(NSM Q(PSF)	dynamic pressure; 1/2ρV <sup>2</sup>
F <sub>N</sub>		normal force
FA		axial force
My		pitching moment
Fγ		side force
MZ		yawing moment
Mχ		rolling moment
GAP	GAP	distance between fuselage and fin root chord, in.
SUBSCRIPTS		
b		base
1		local
S		static conditions
t		total conditions
		free stream

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49				0	2 (	0.7	3	213			214			215			
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51						1.0	3	209		208	207						
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FIGURE 1. AXIS SYSTEMS AND POSITIVE SIGN CONVENTION,
BODY AXIS SYSTEM

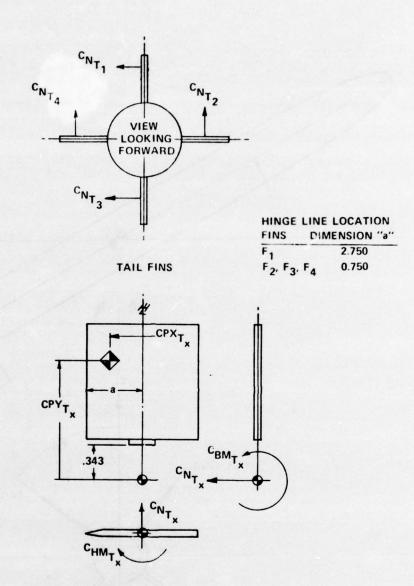


FIGURE 2. TAIL FIN AXIS SYSTEMS , POSITIVE SIGN CONVENTION
AND CENTER OF PRESSURE LOCATION

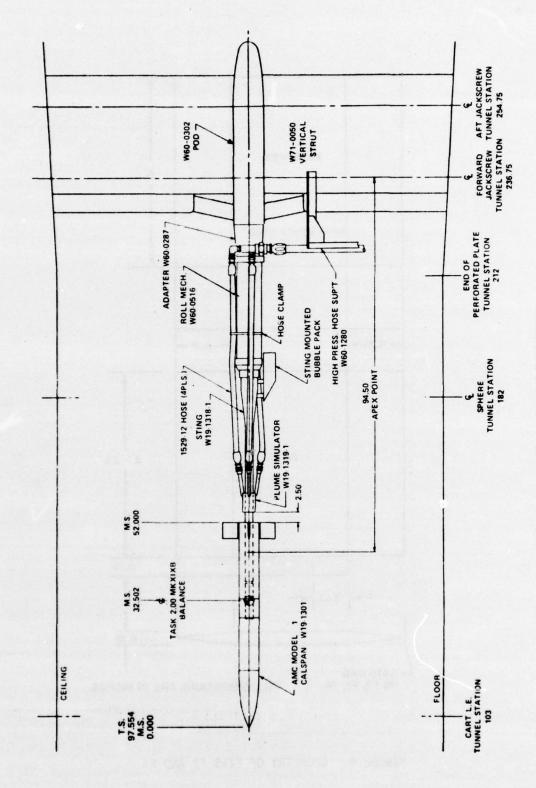


FIGURE 3. INSTALLATION DRAWING

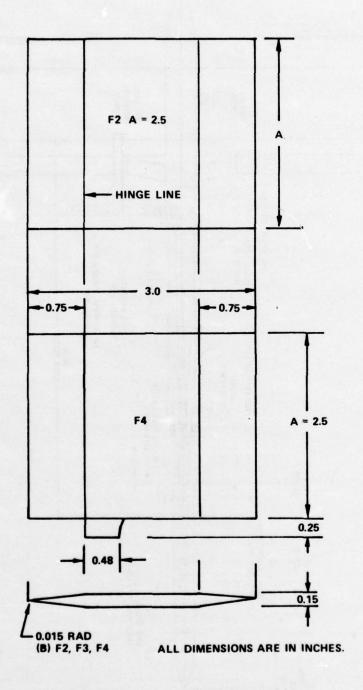


FIGURE 4. GEOMETRY OF FINS F2 AND F4

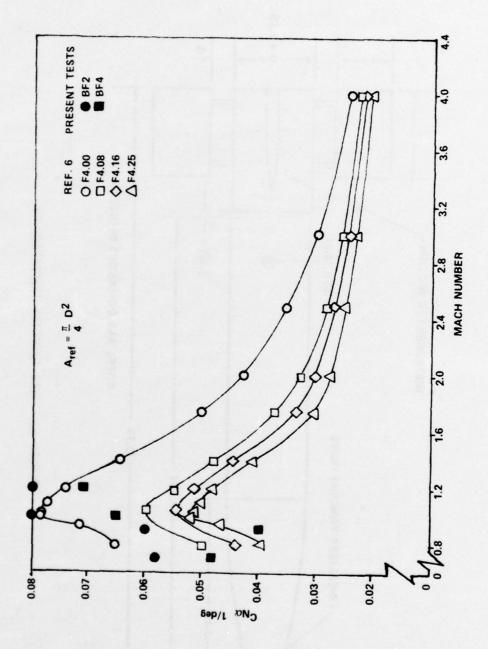


FIGURE 5. NORMAL FORCE COEFFICIENT SLOPE VERSUS MACH NUMBER

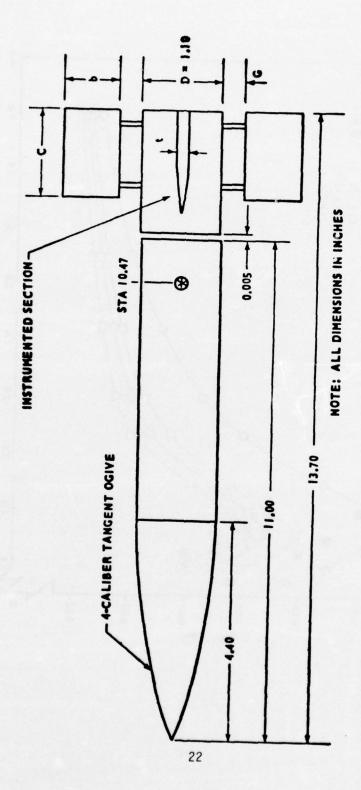


FIGURE 6. MODEL GEOMETRY (REFERENCE 6)

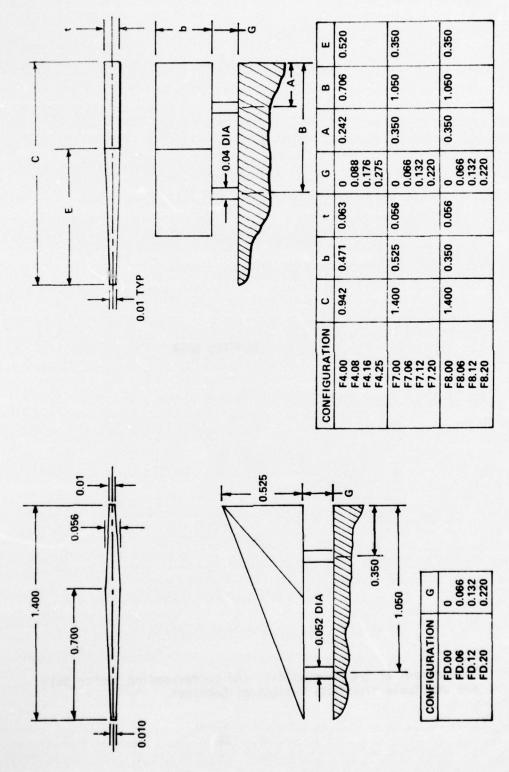
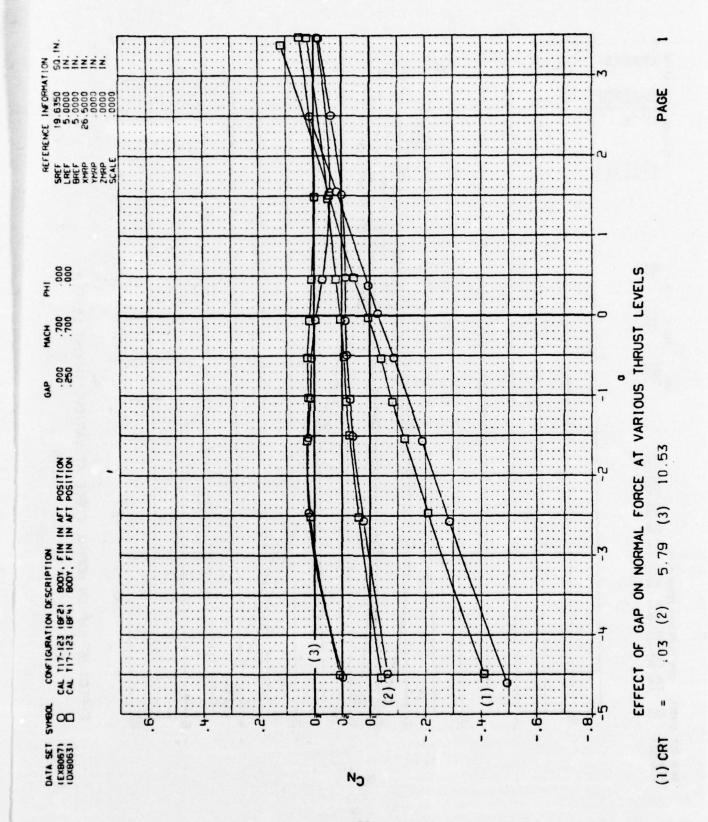
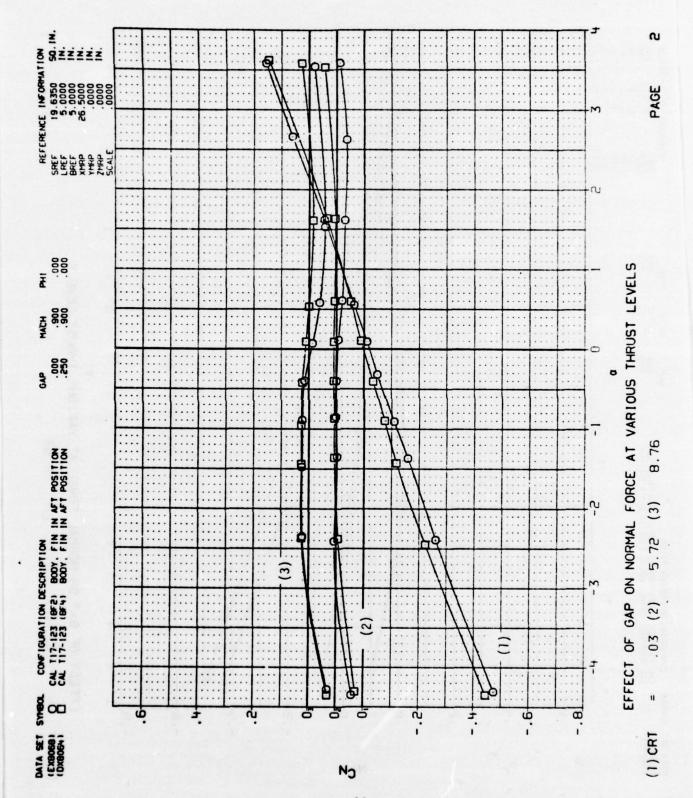


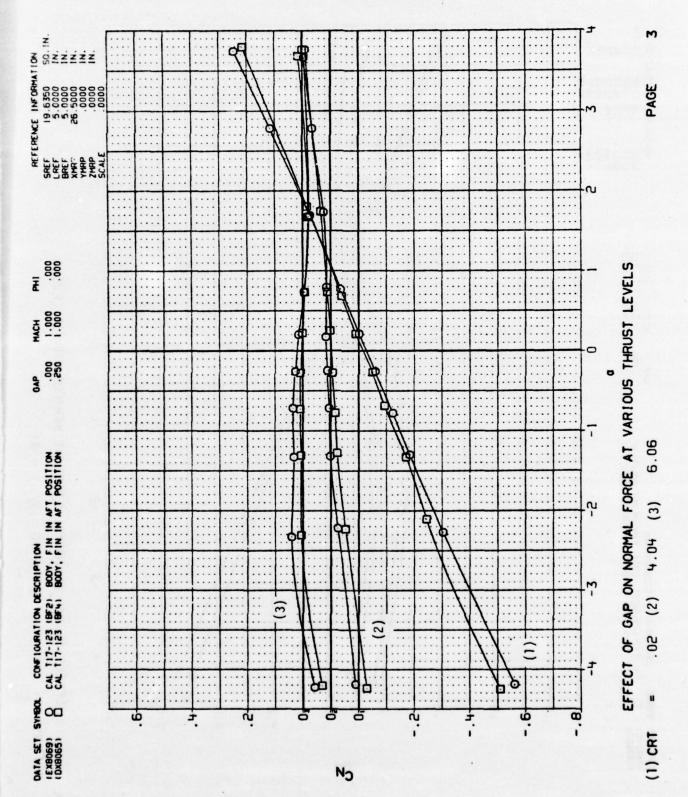
FIGURE 7. FIN GEOMETRY (REFERENCE 6)

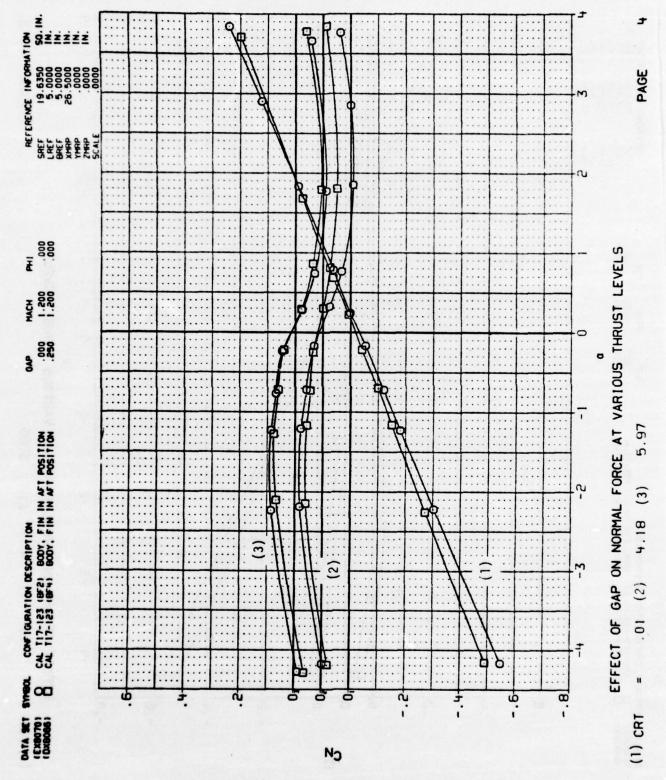
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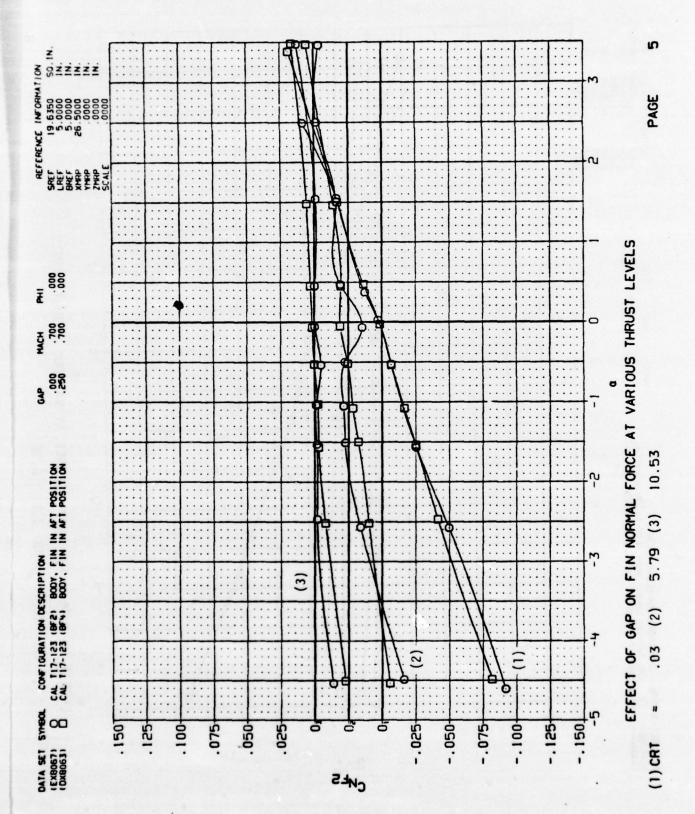
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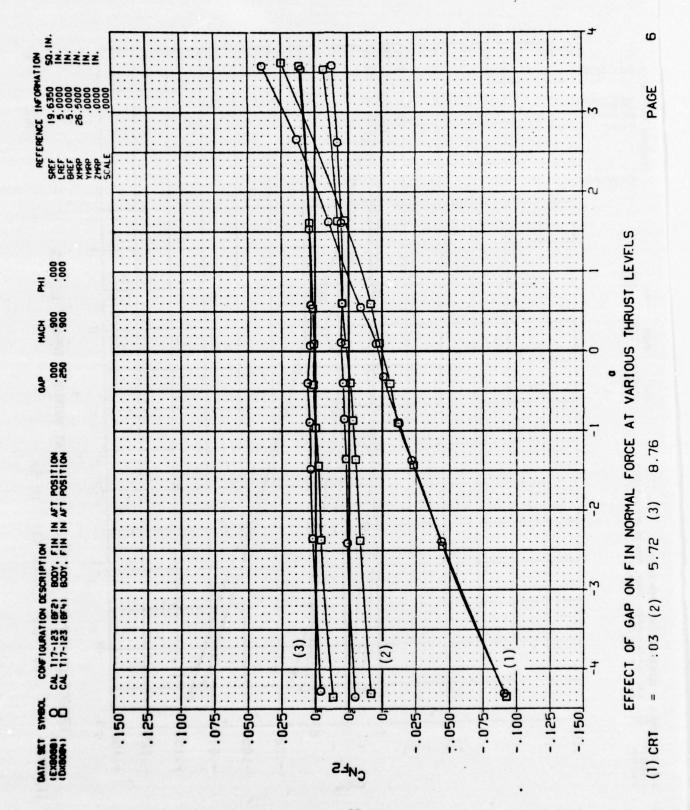


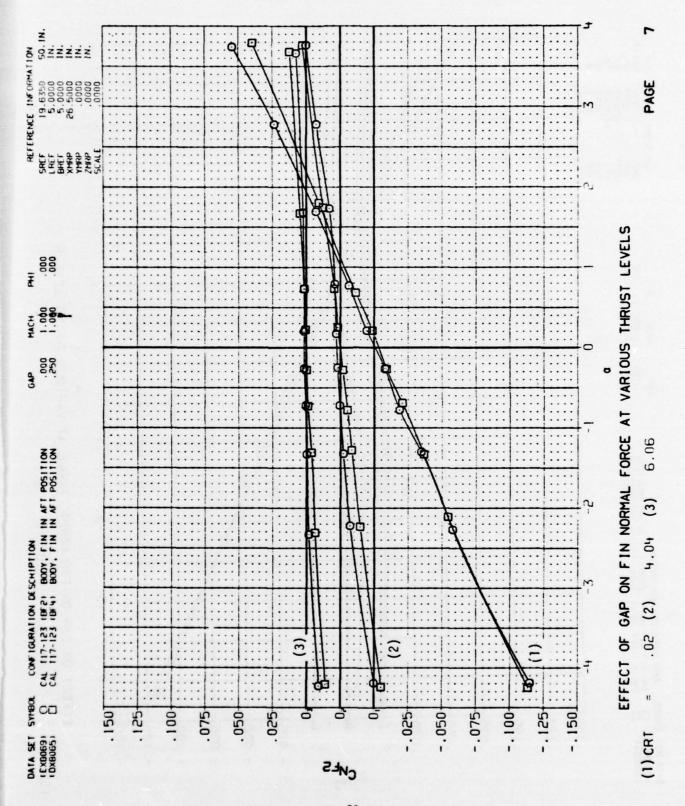


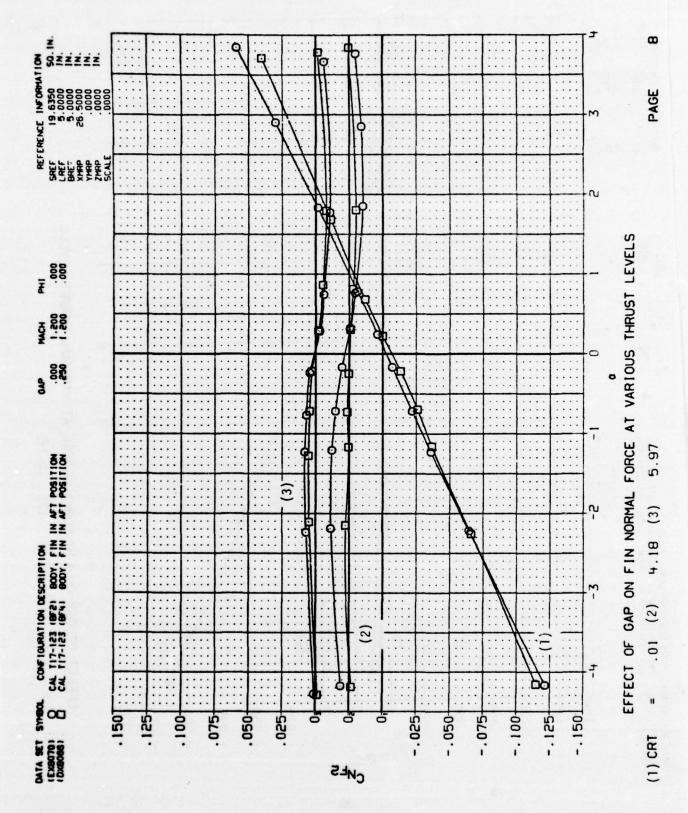


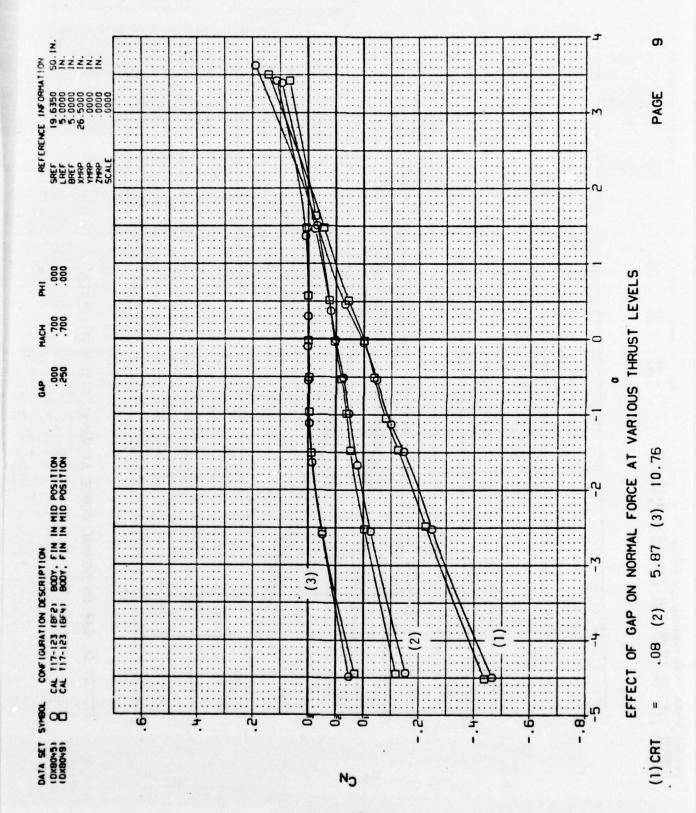


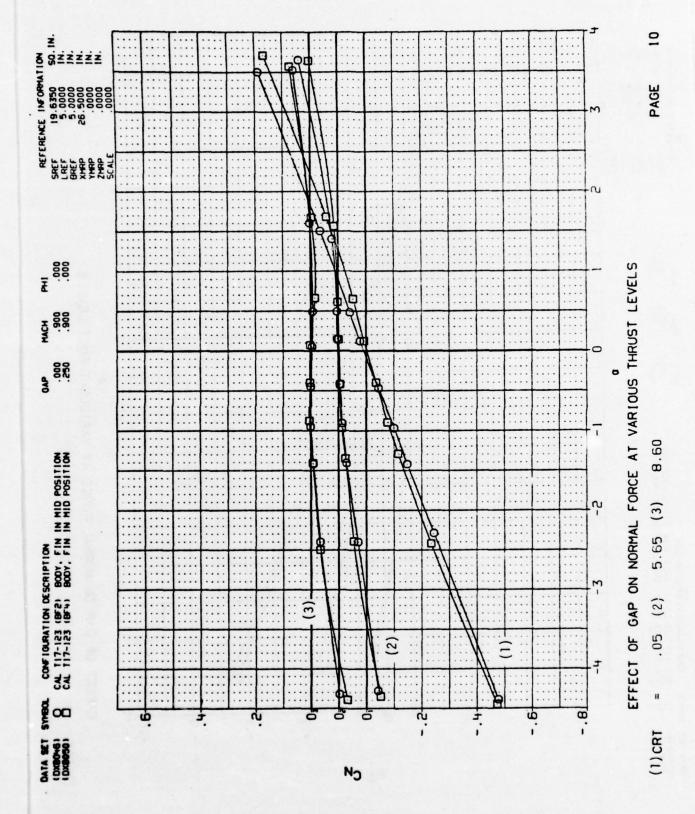


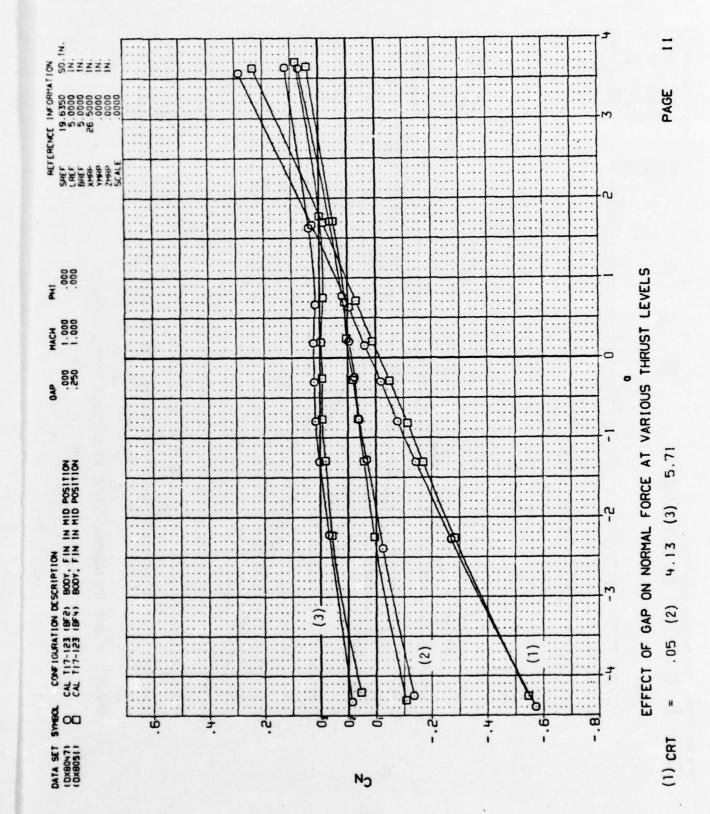


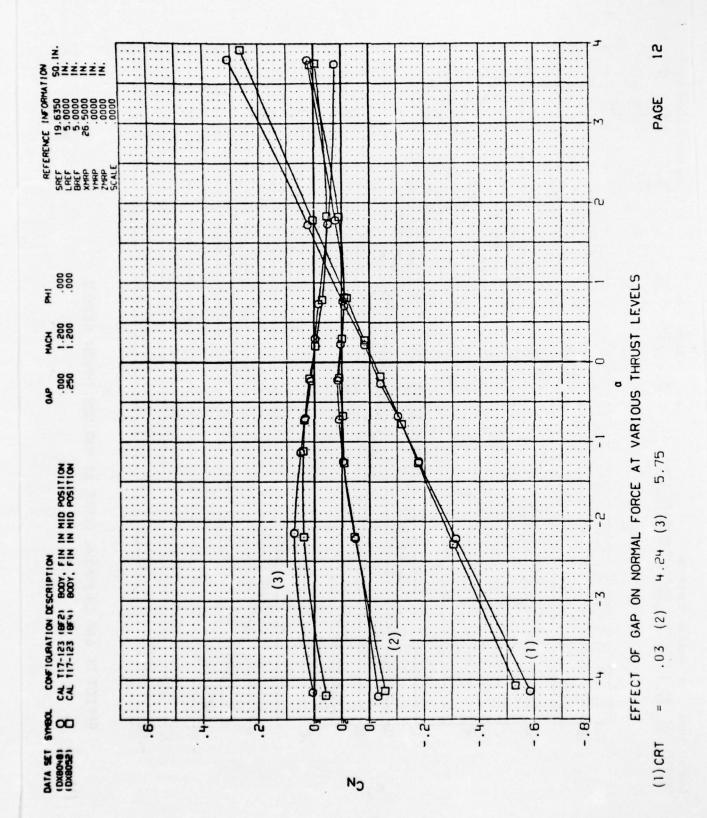


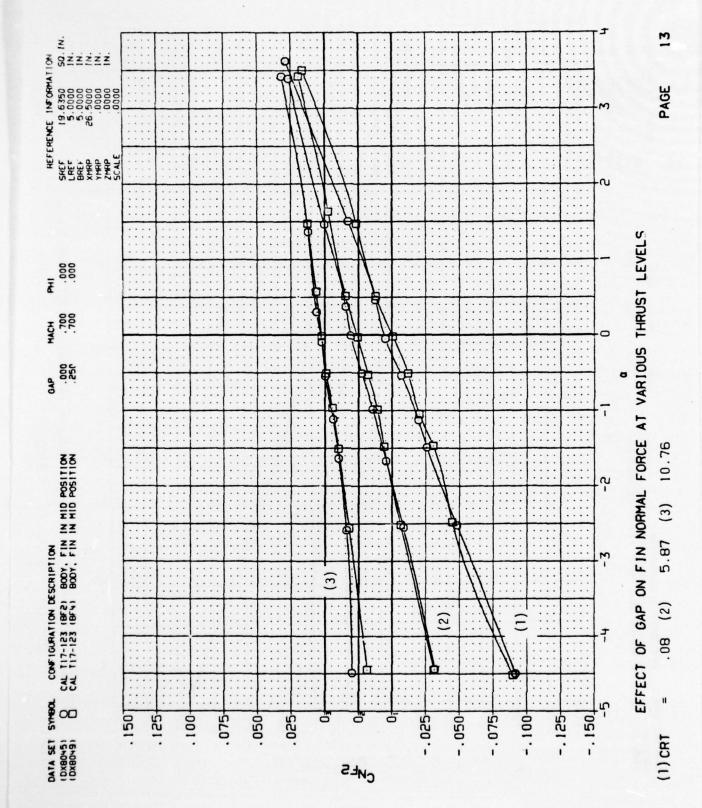


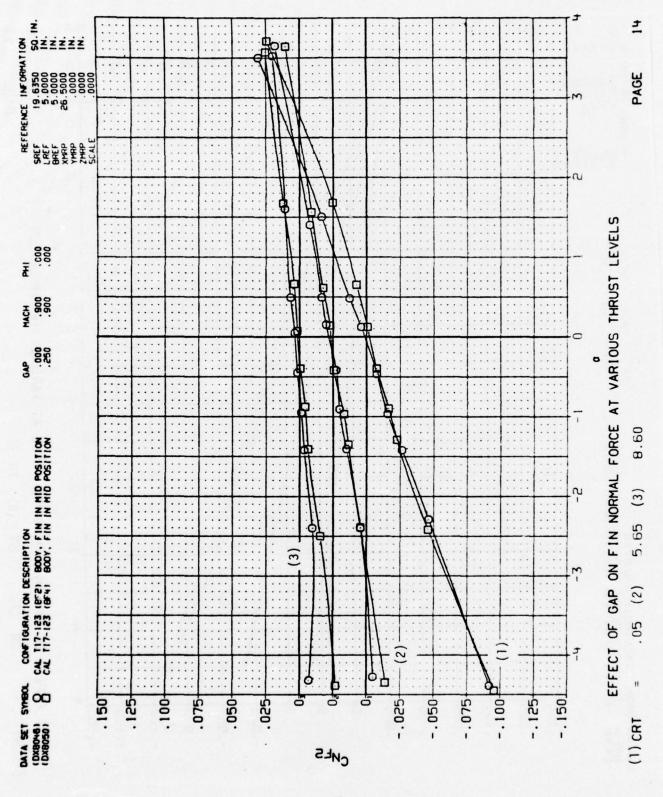


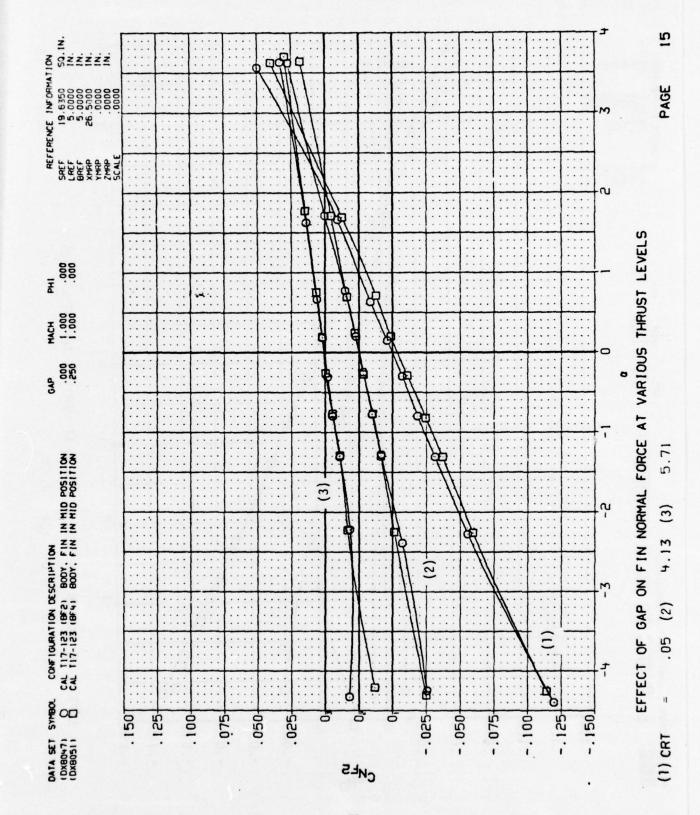


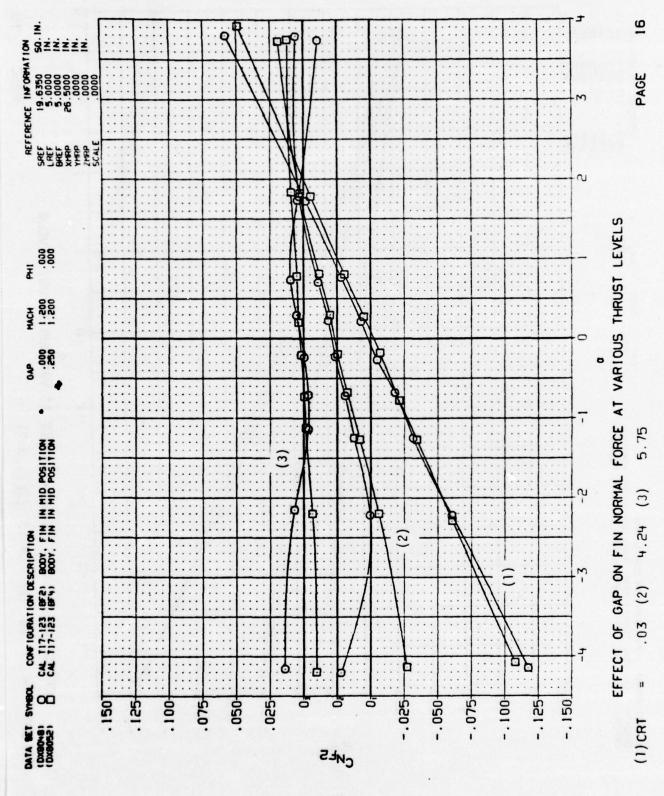


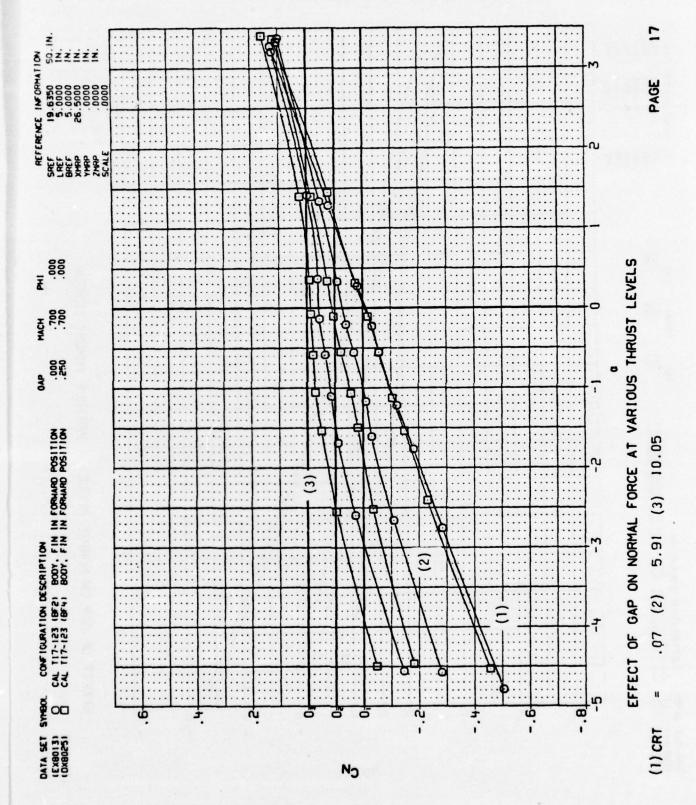


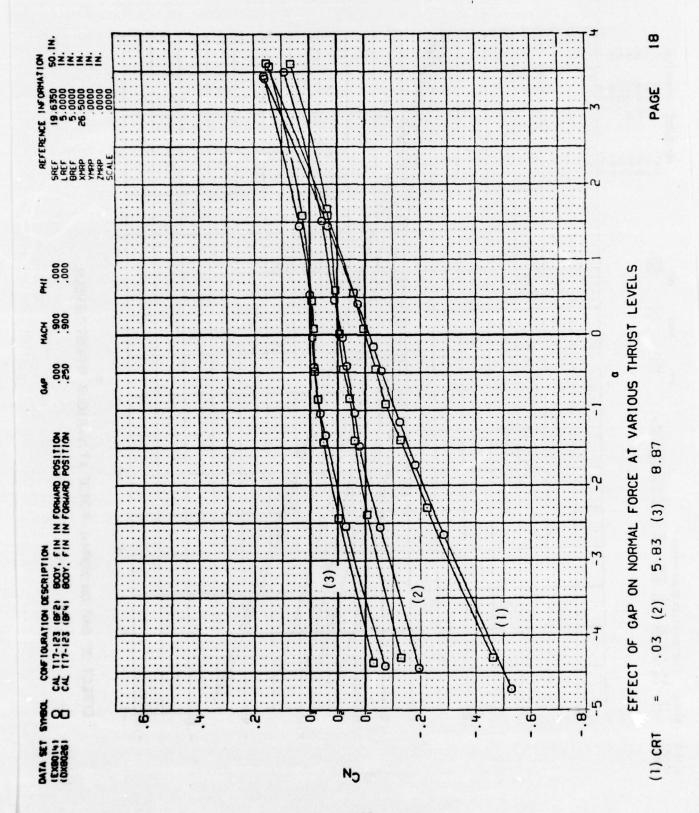


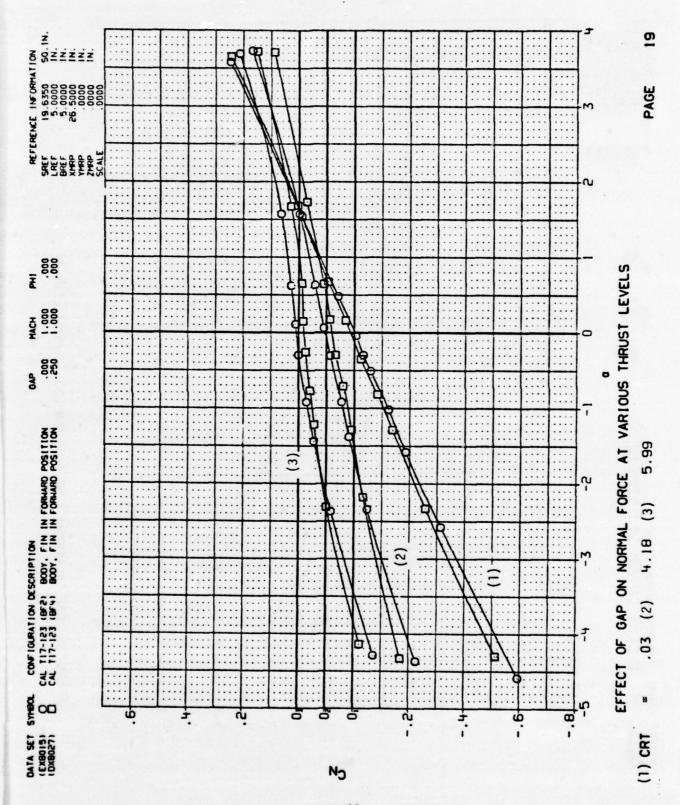


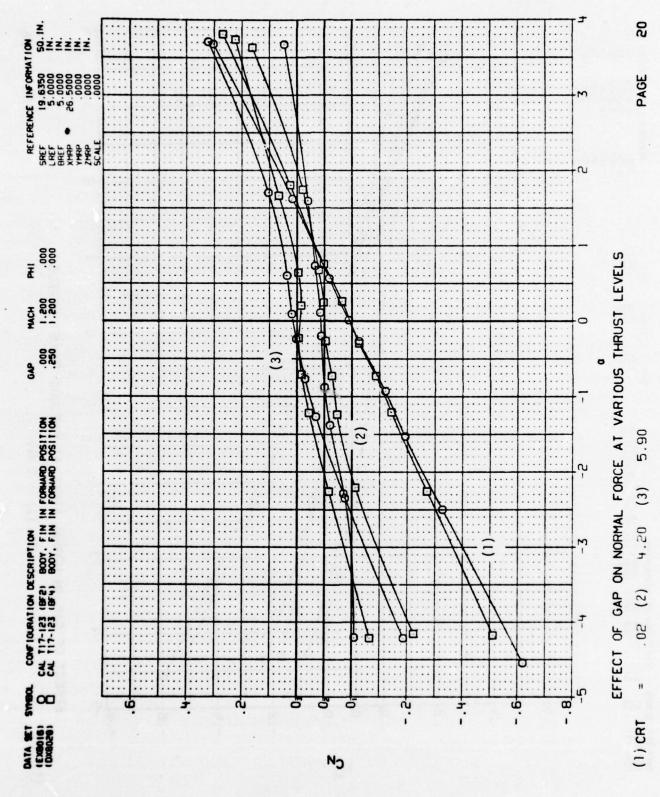


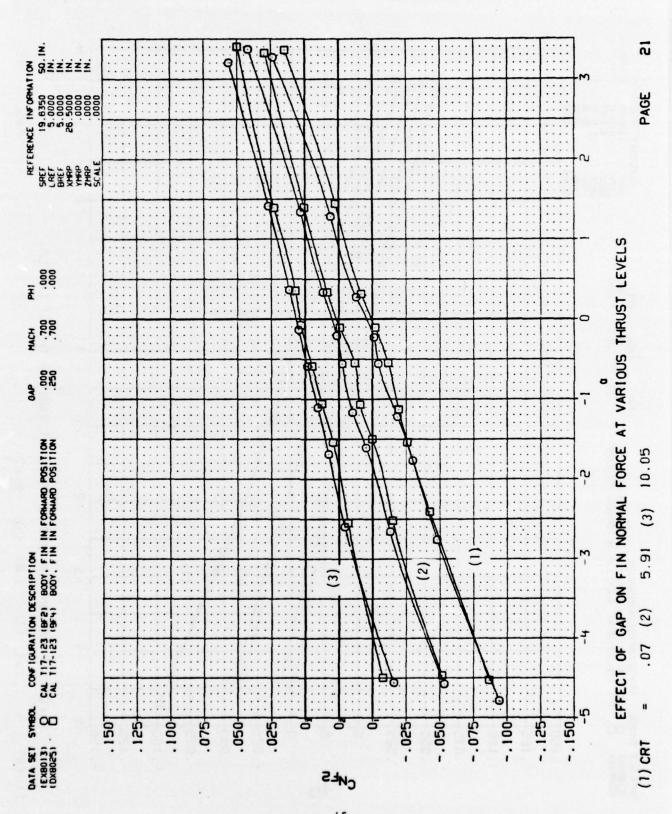


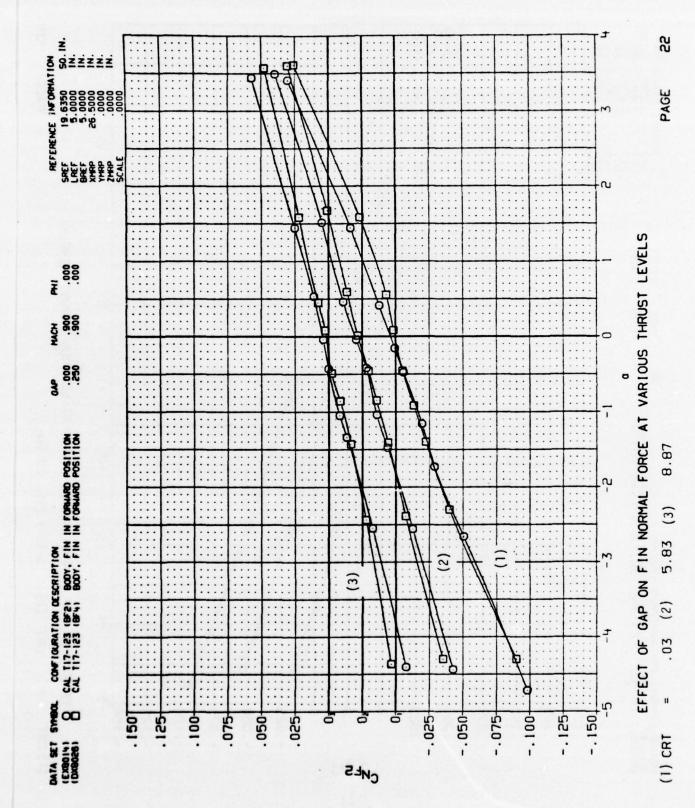


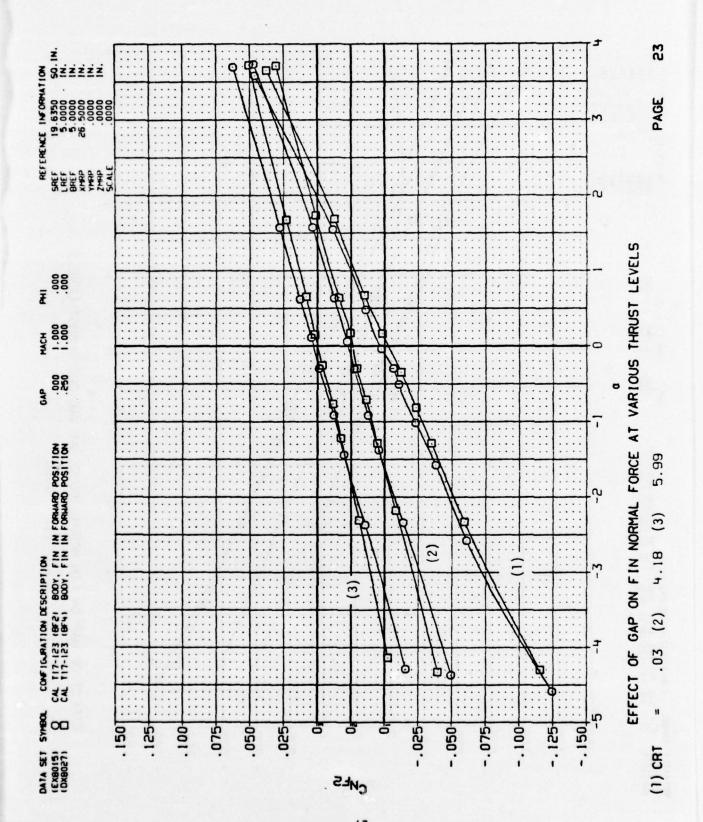


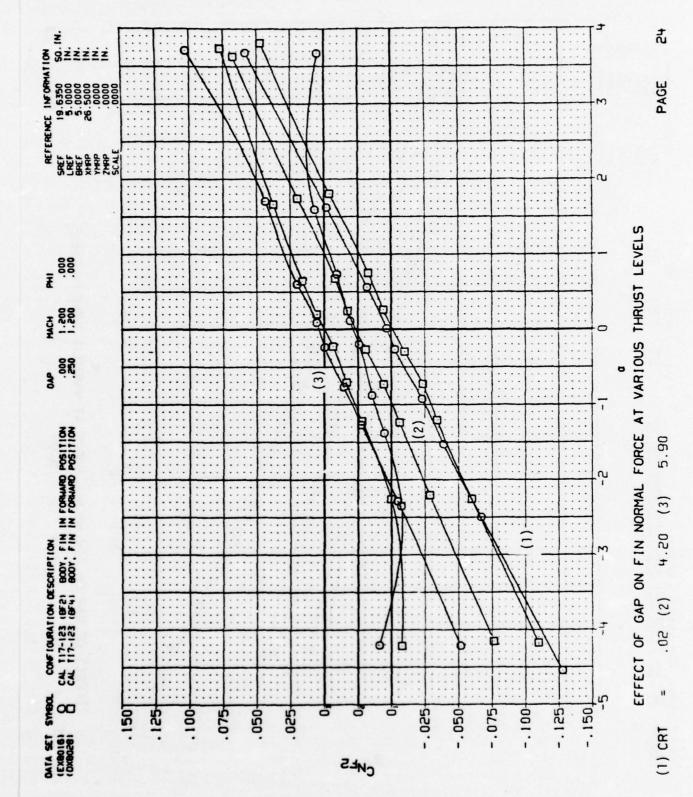


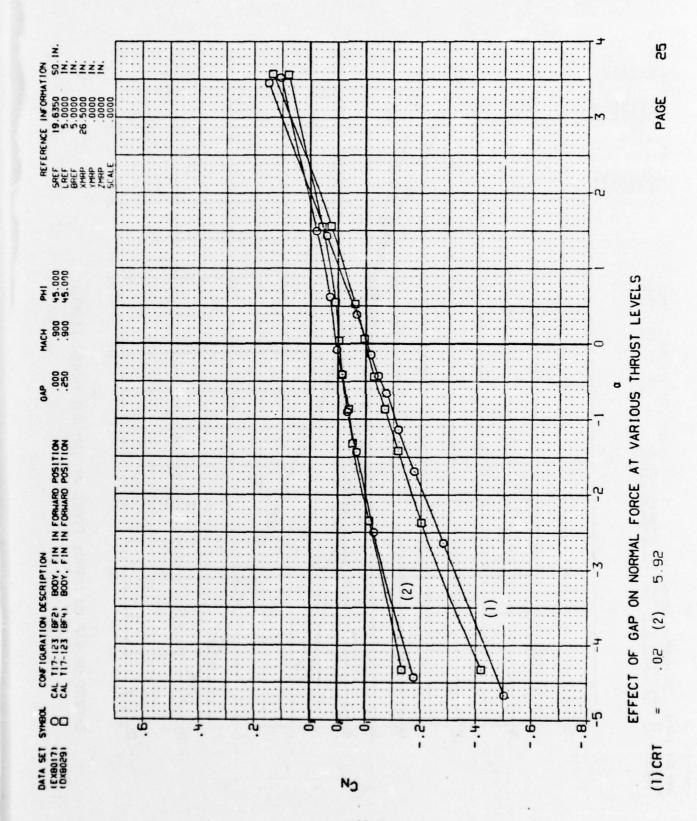


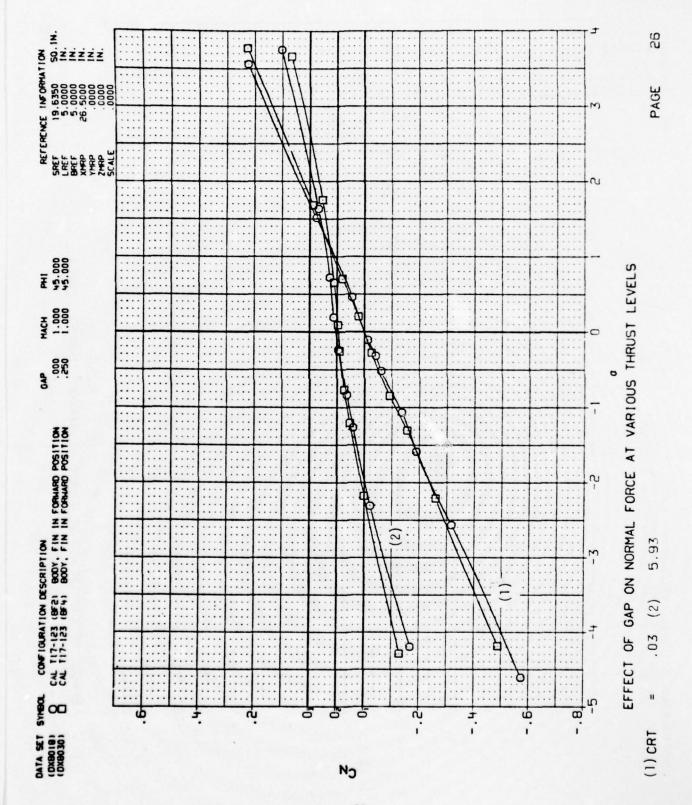


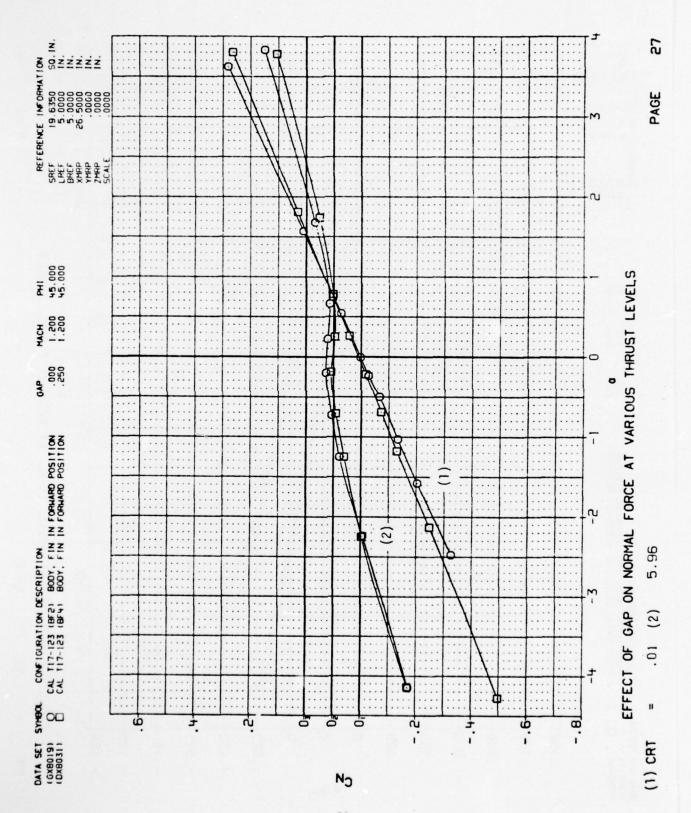


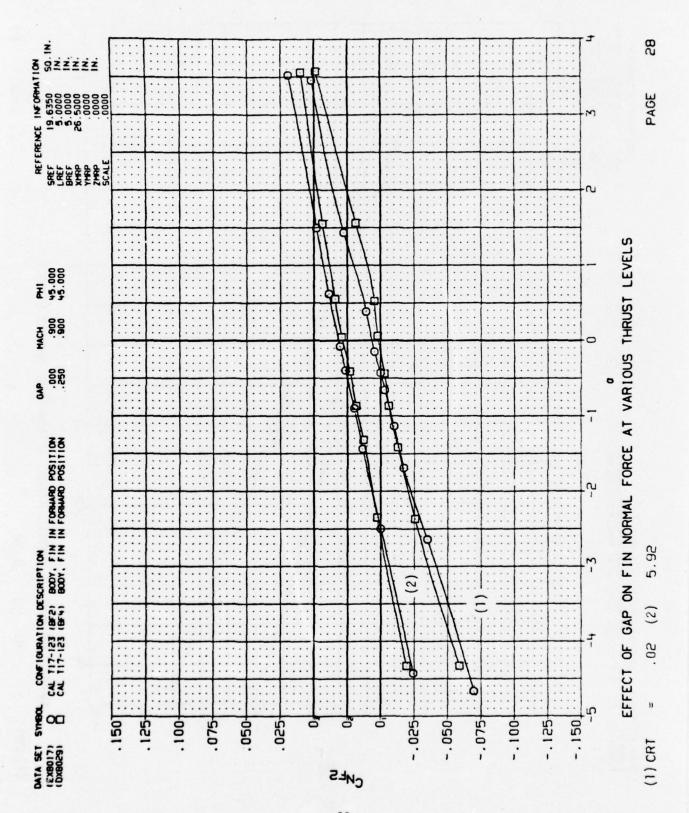


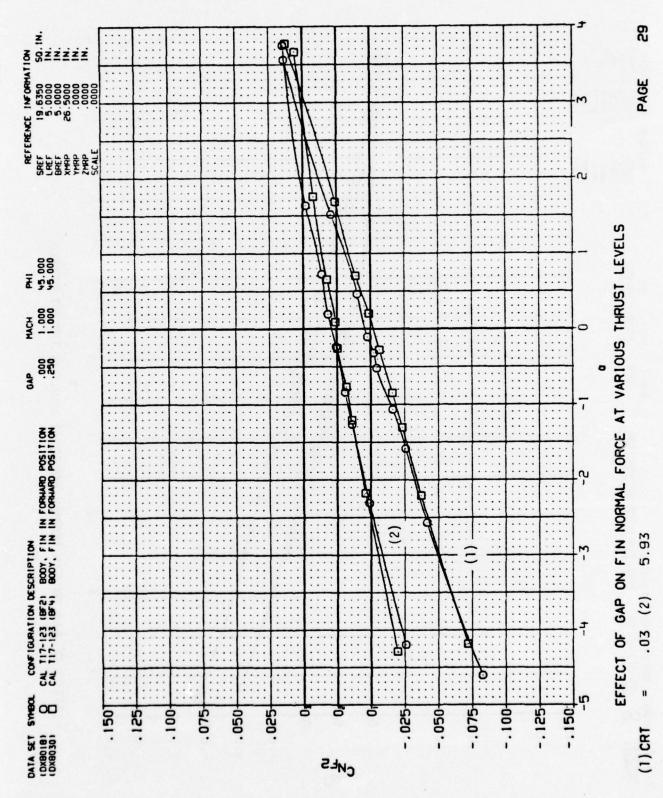


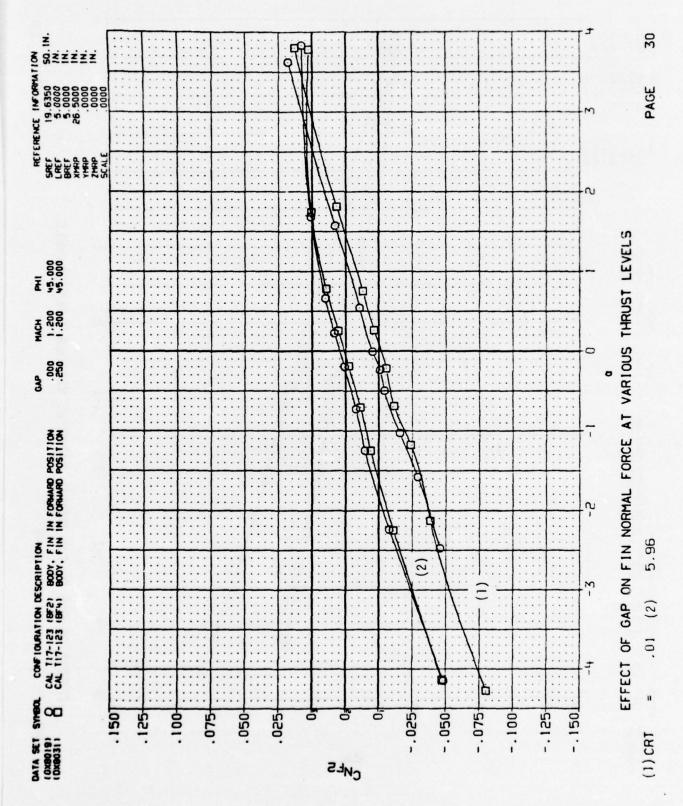


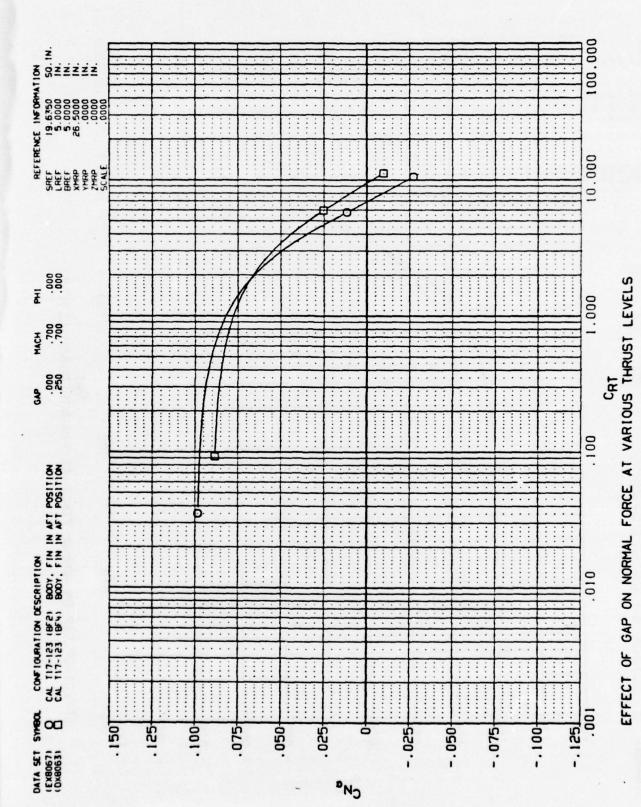


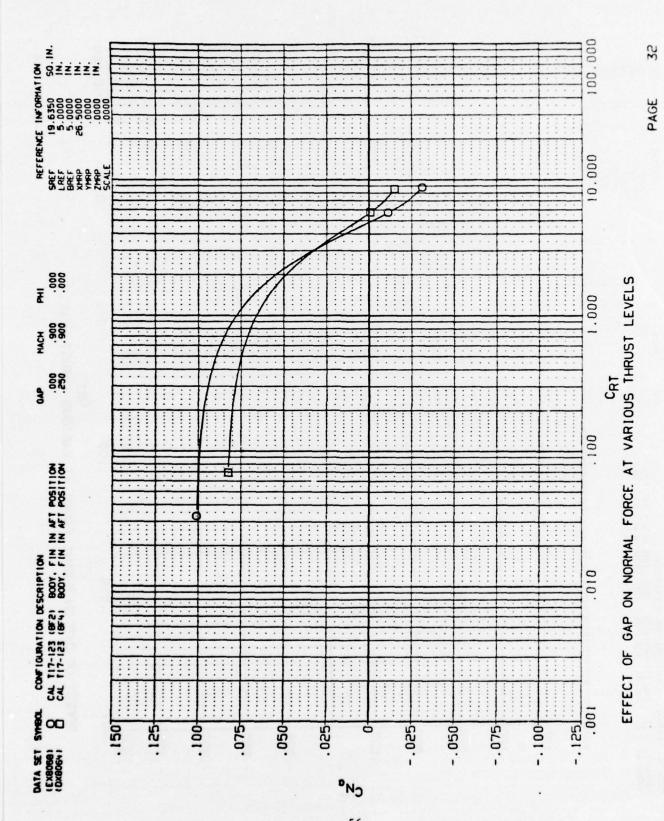


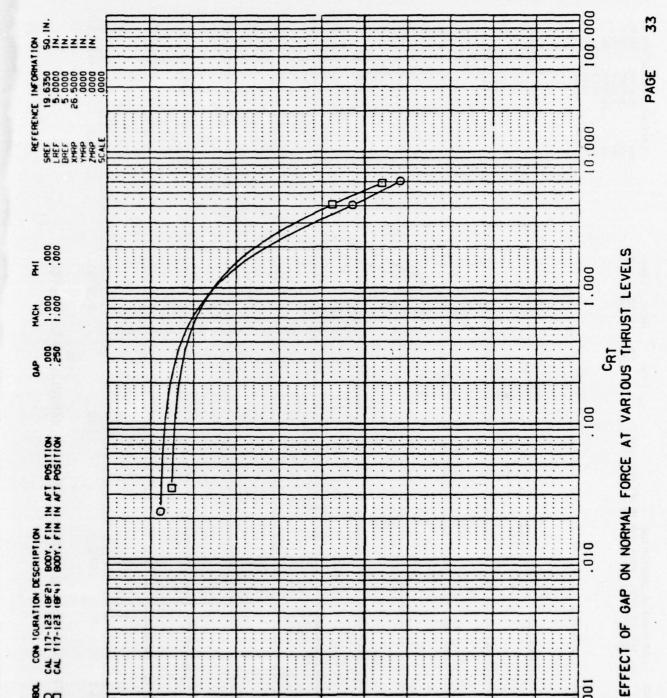












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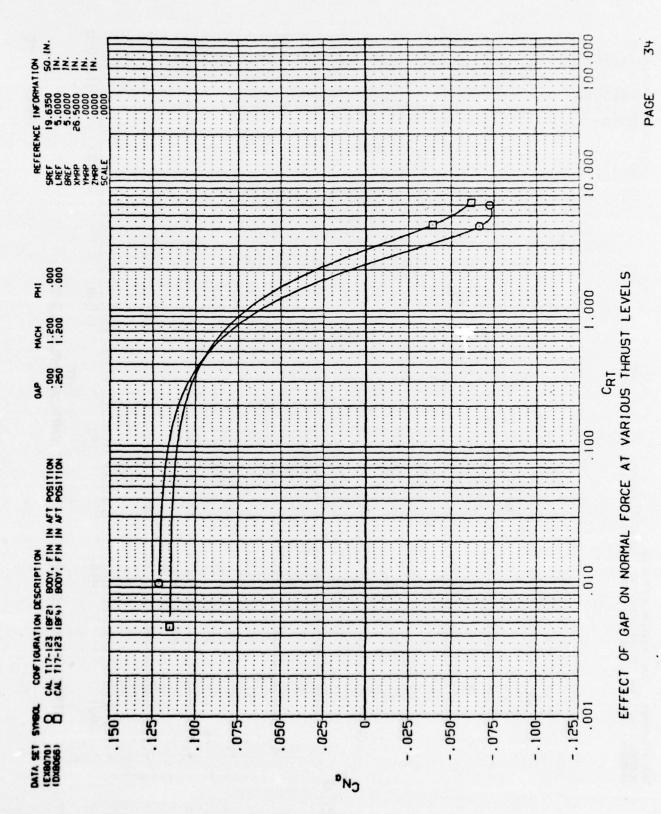
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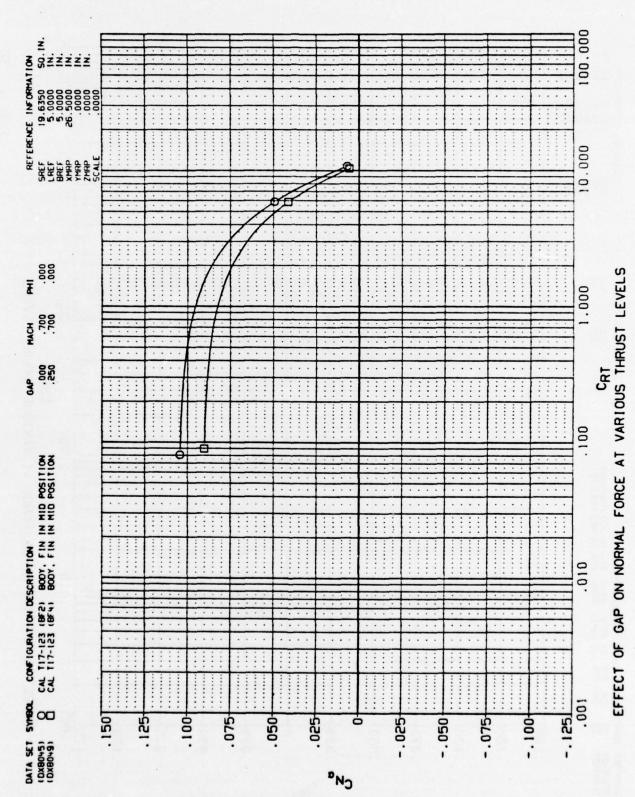
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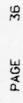
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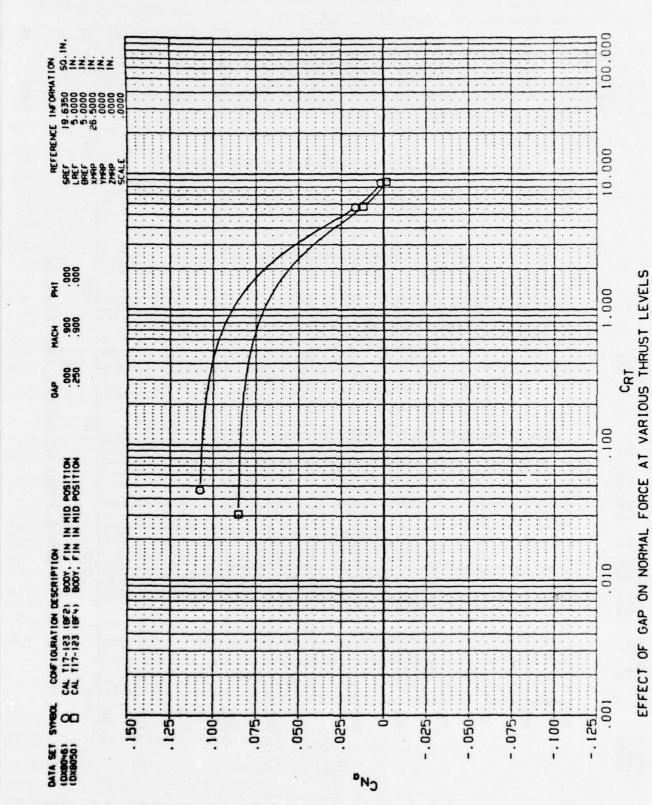
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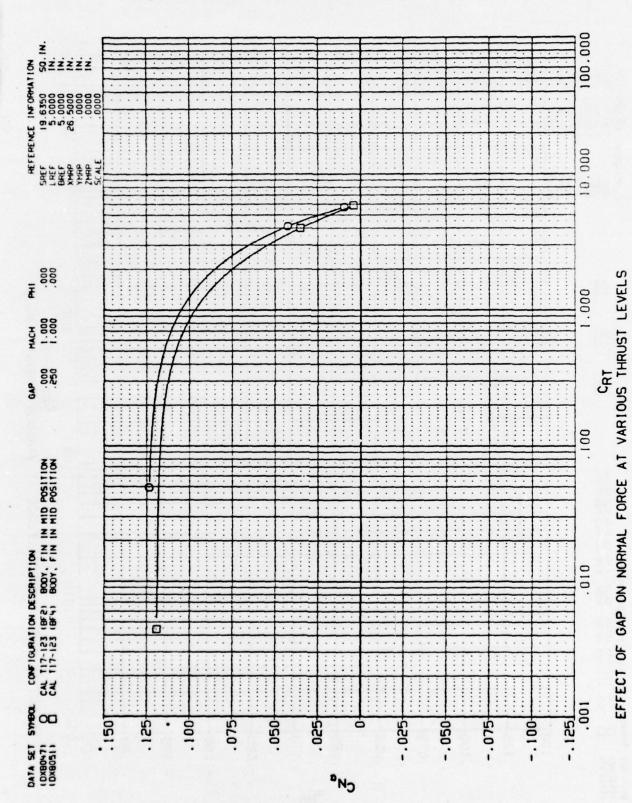


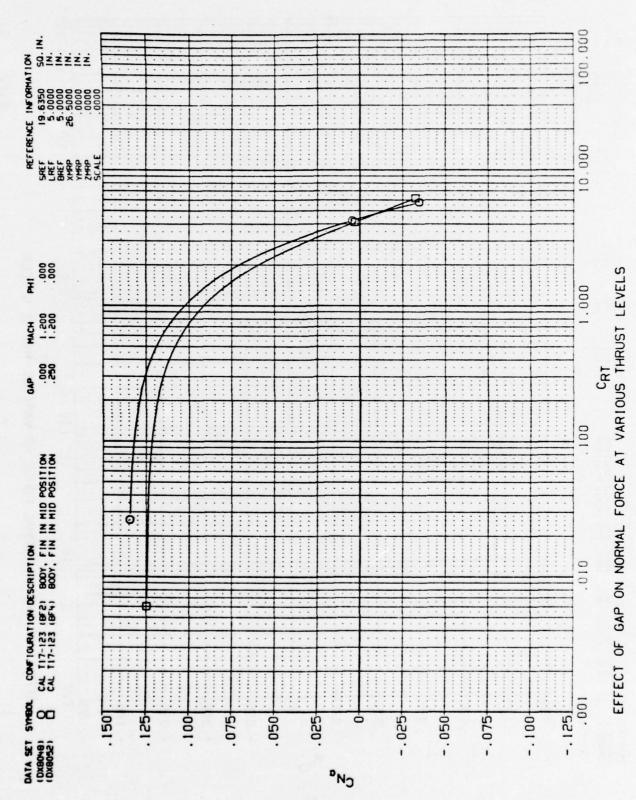


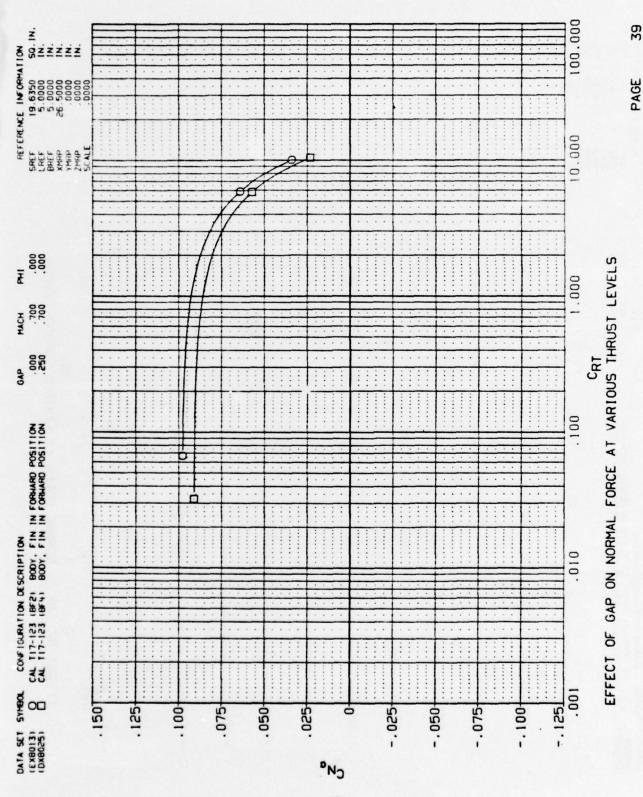


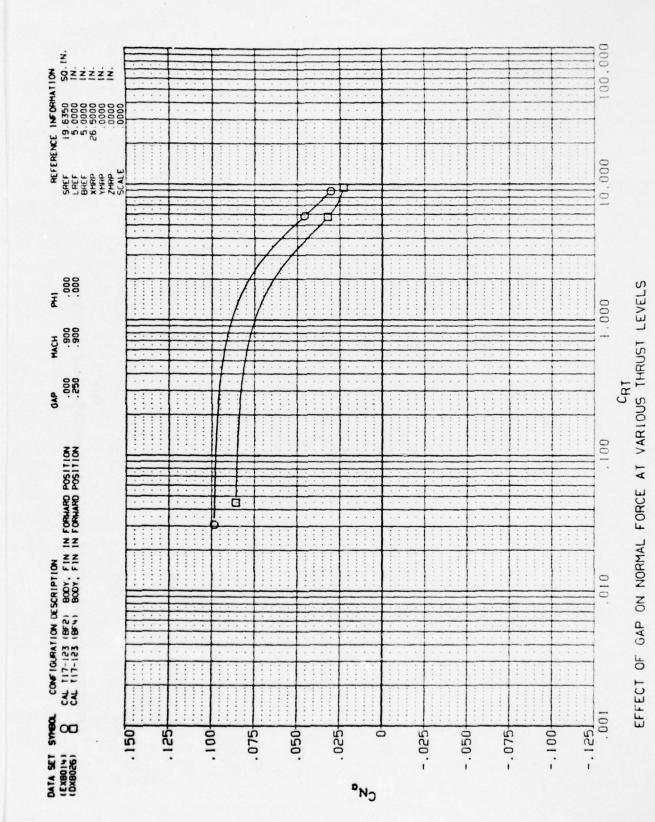


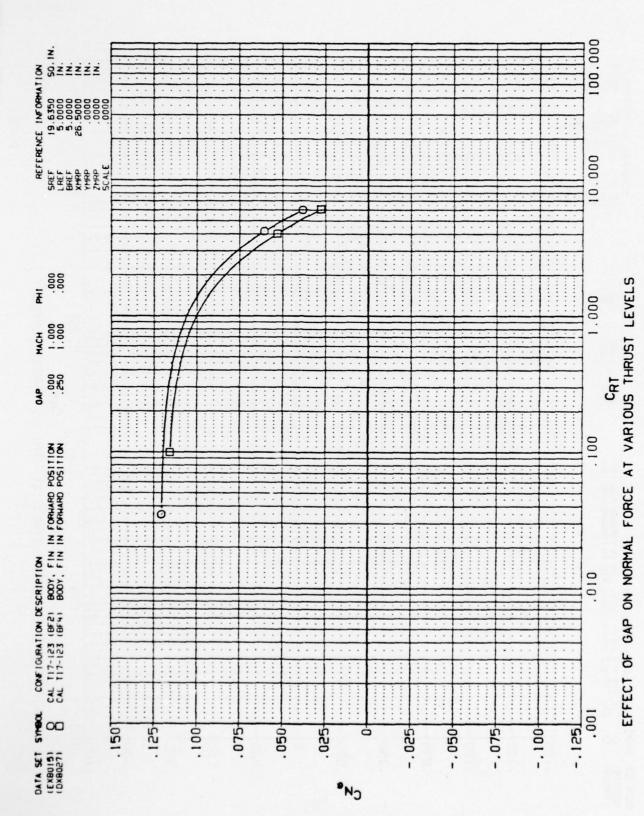


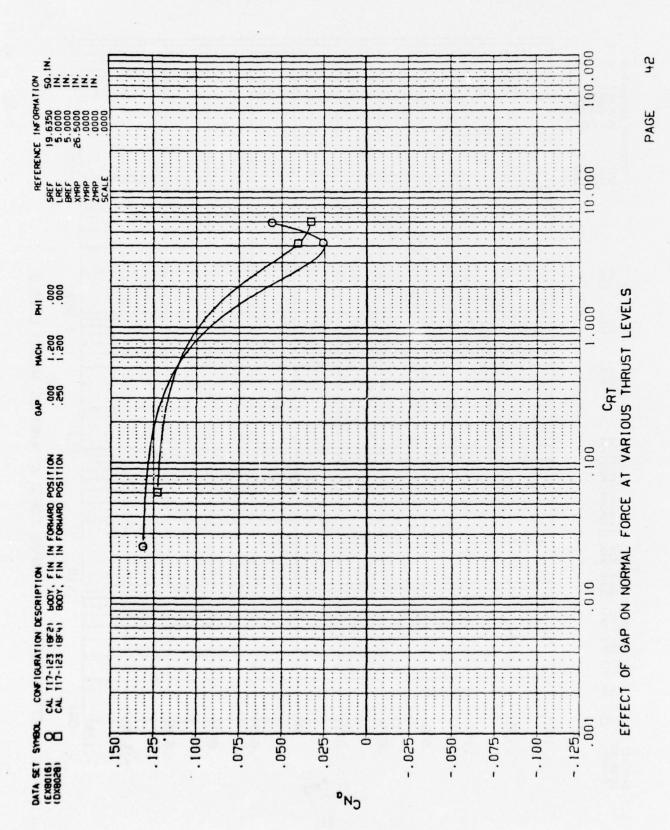


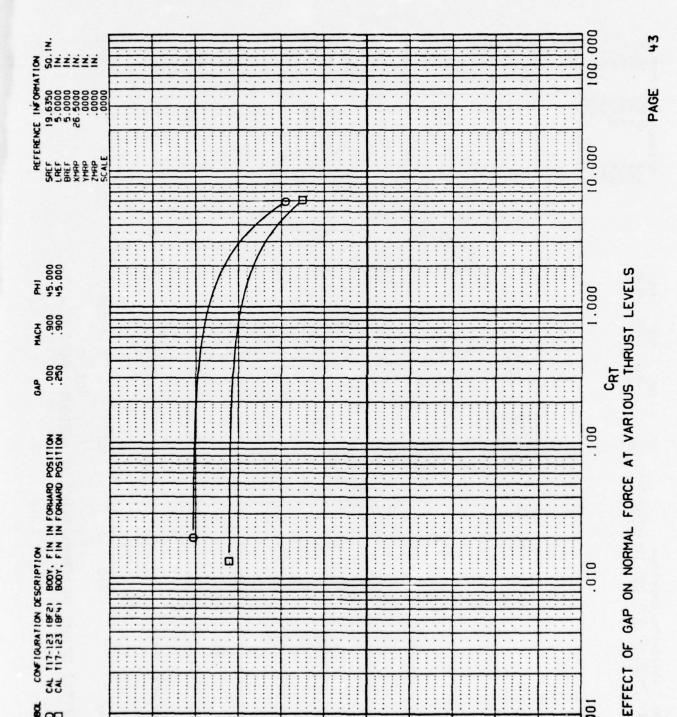












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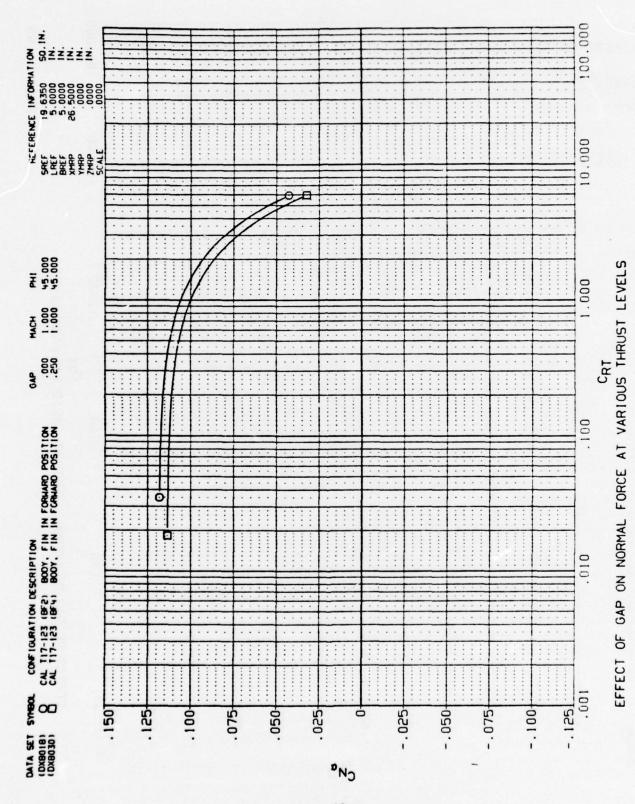
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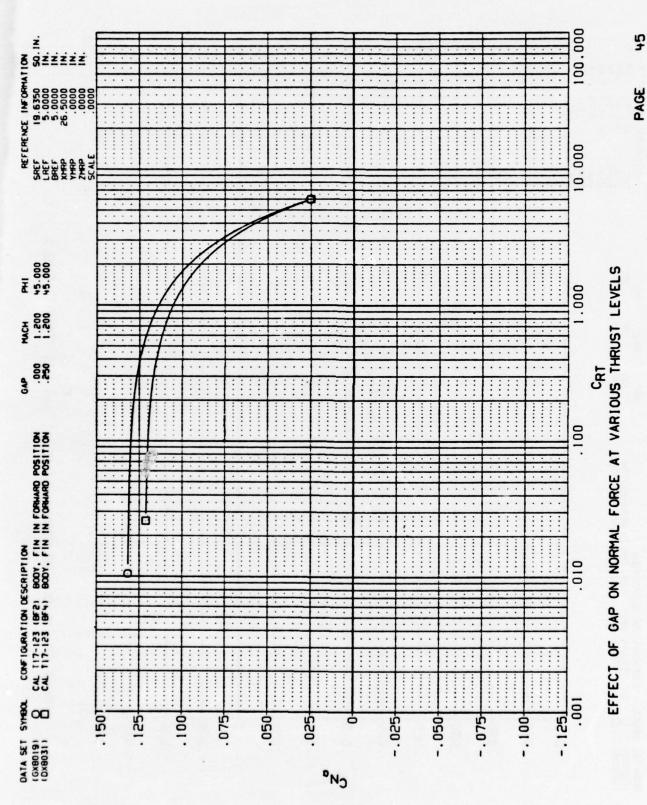
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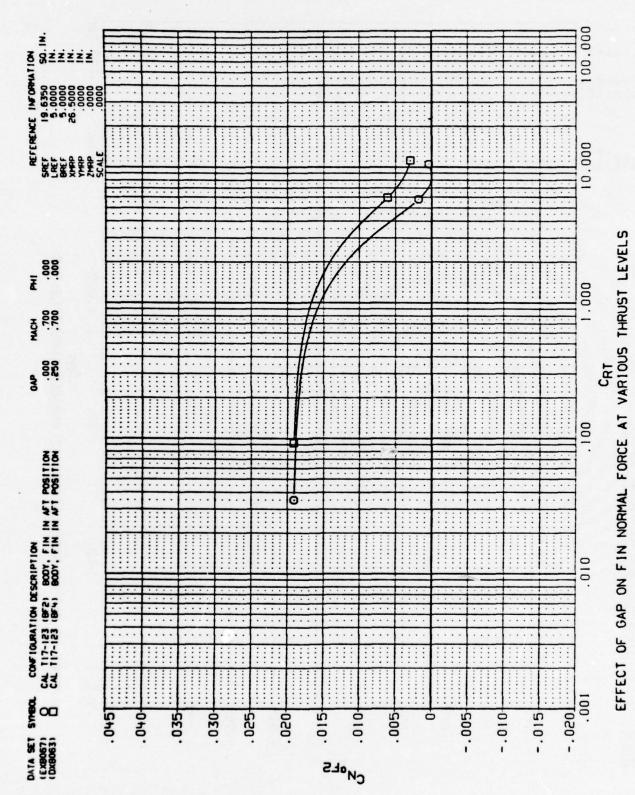
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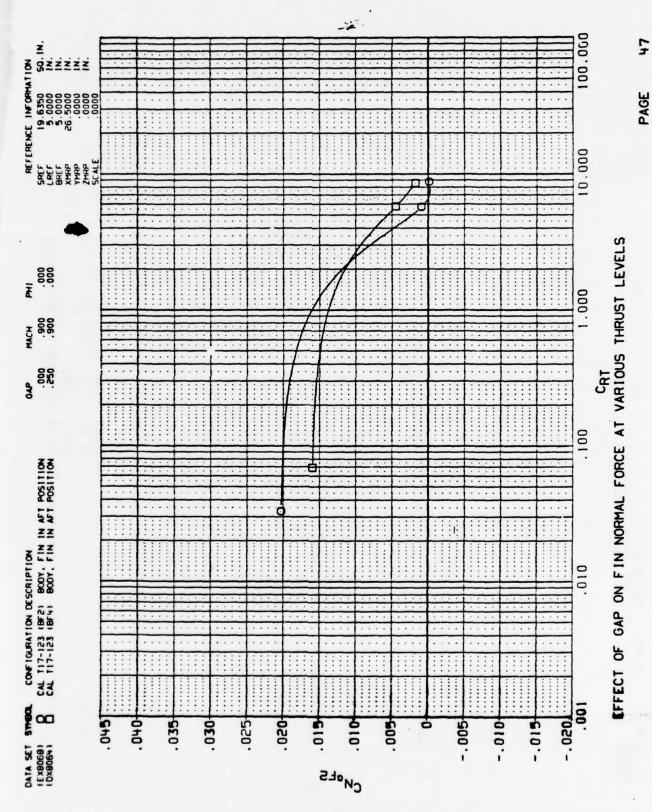
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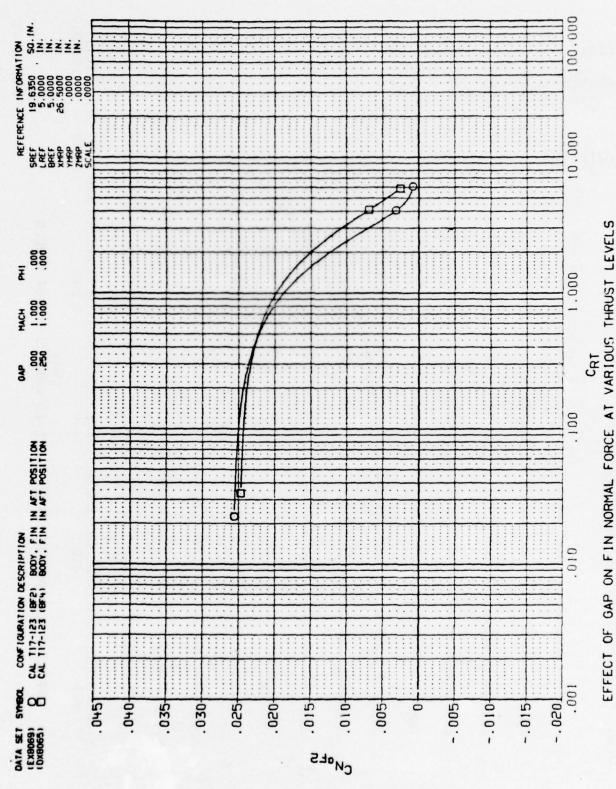


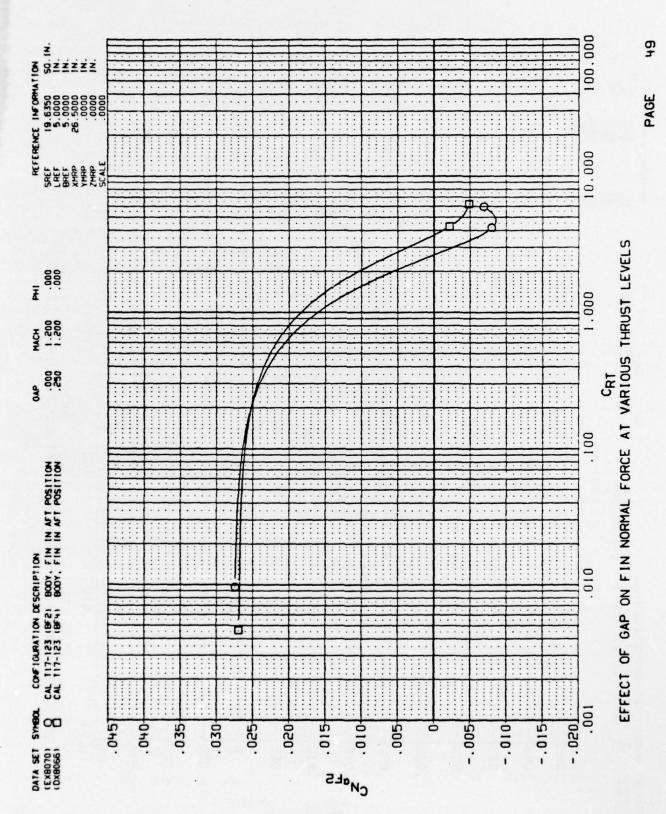
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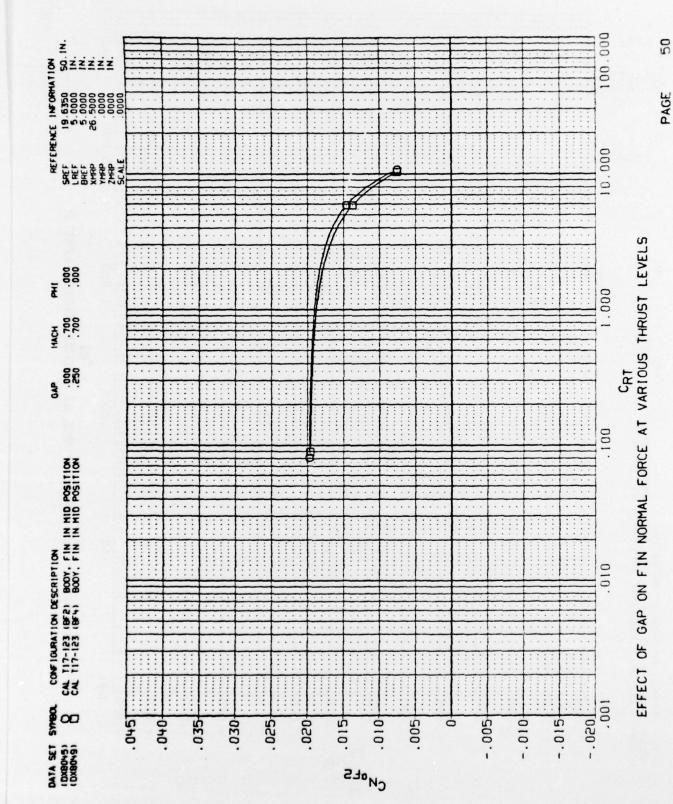




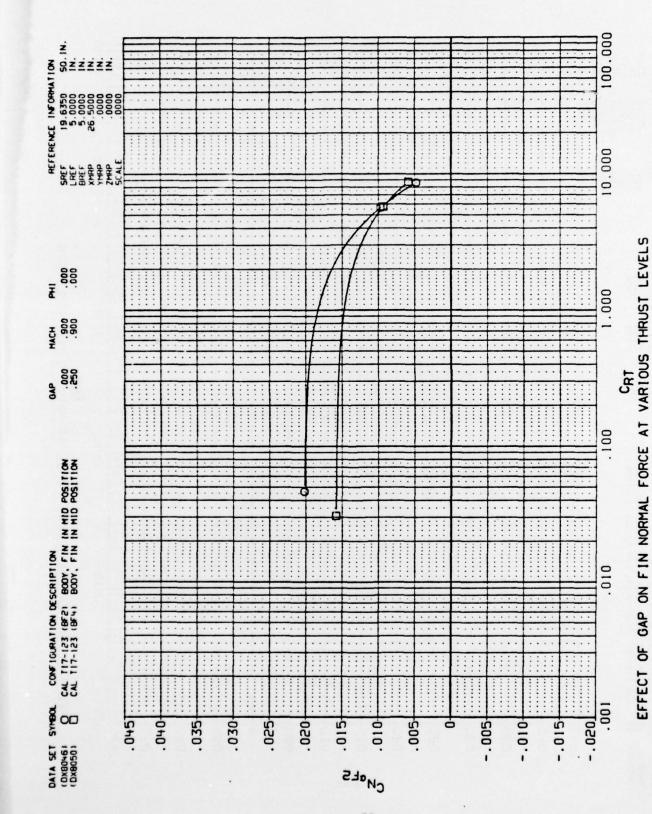


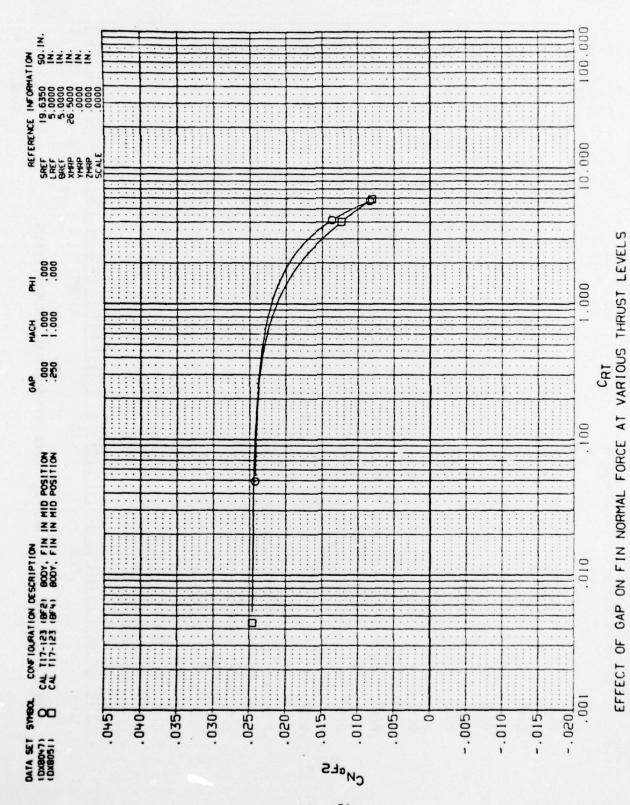


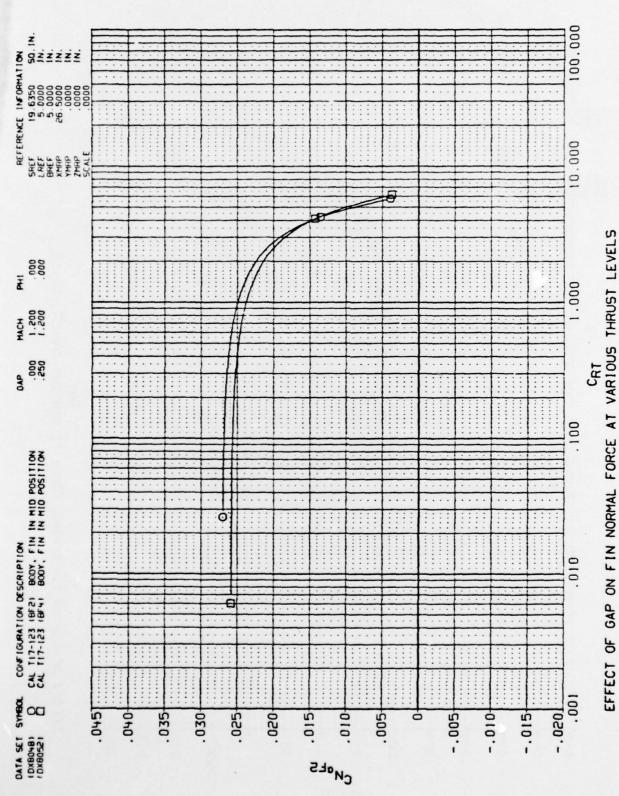


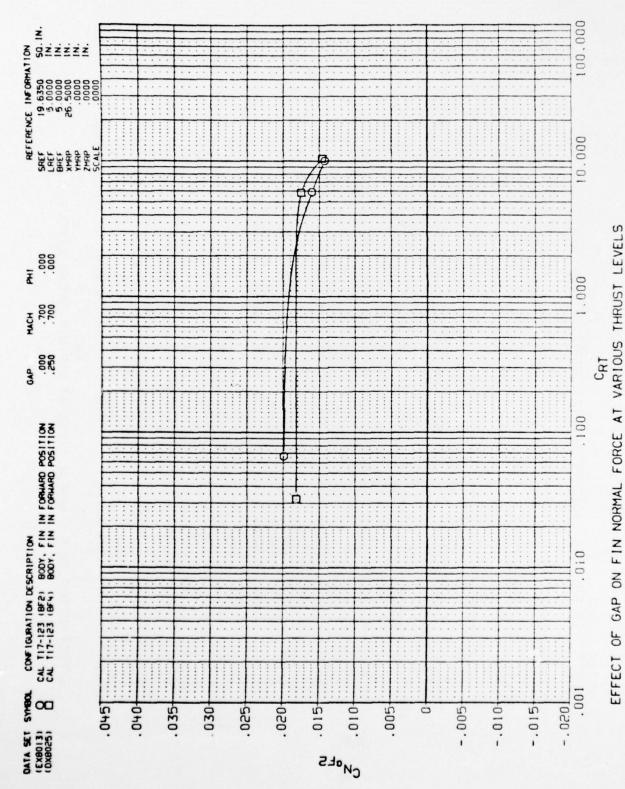


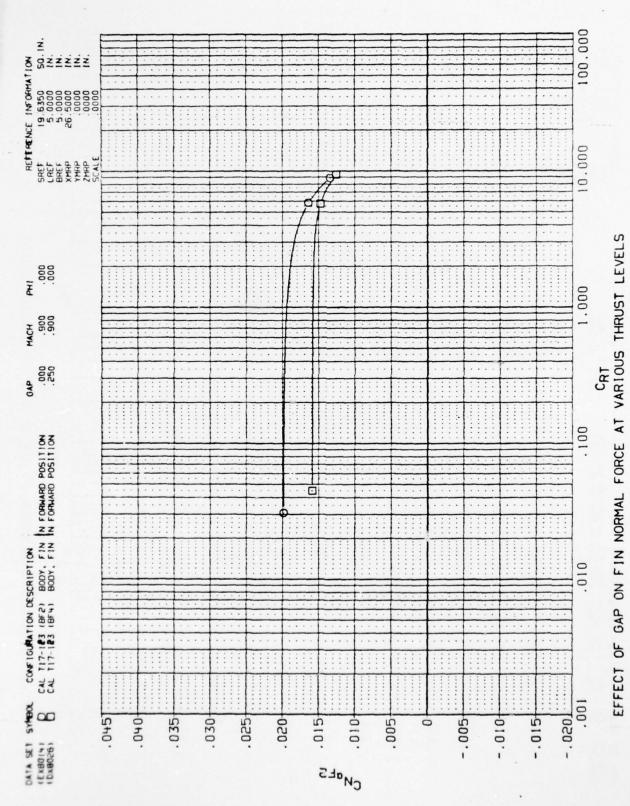


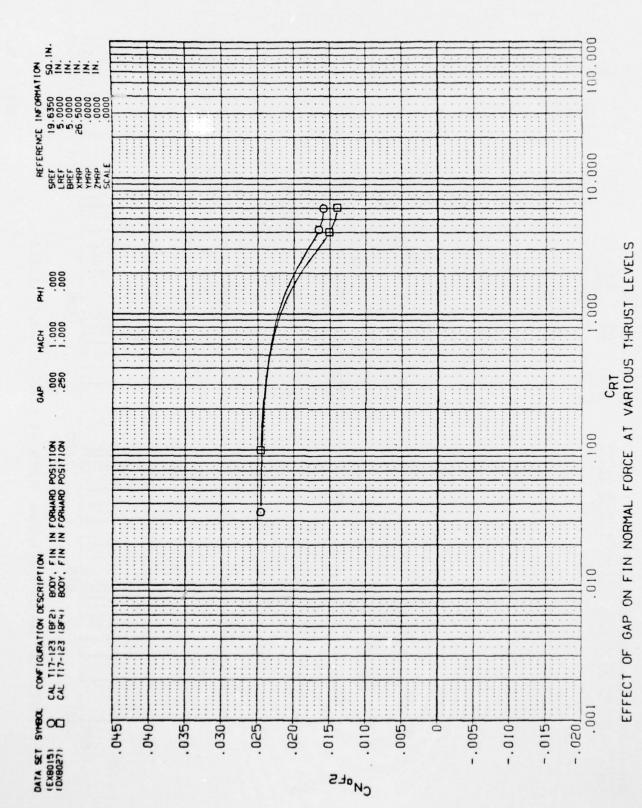


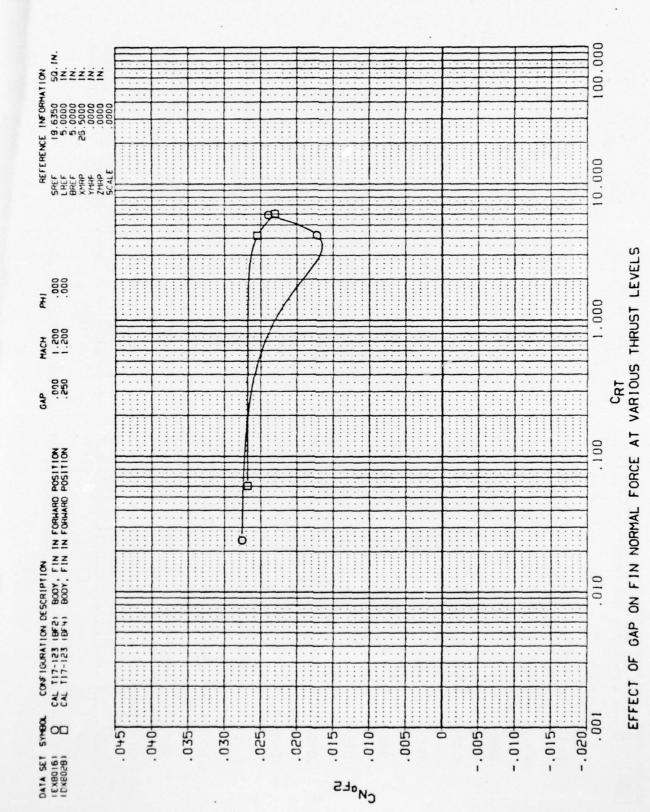


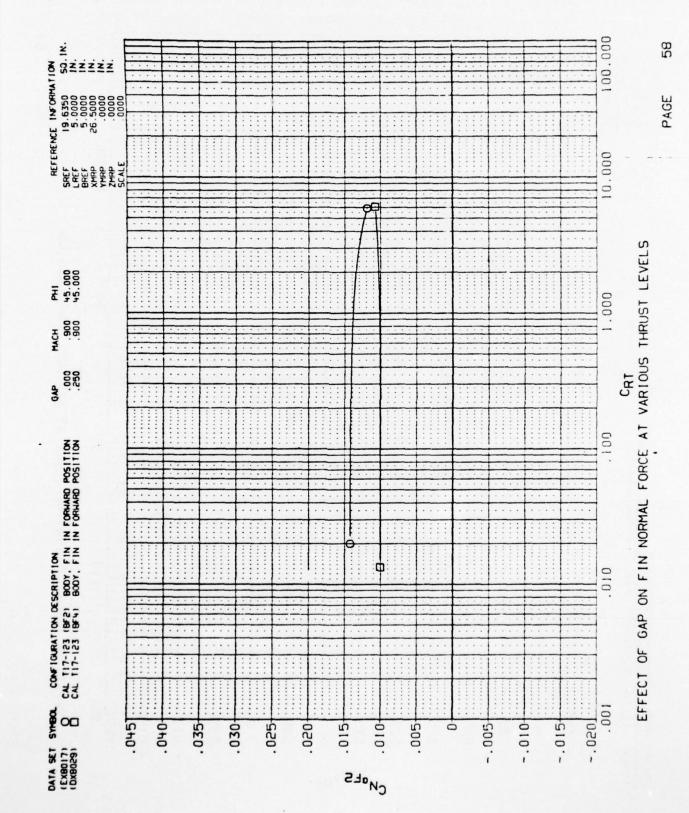


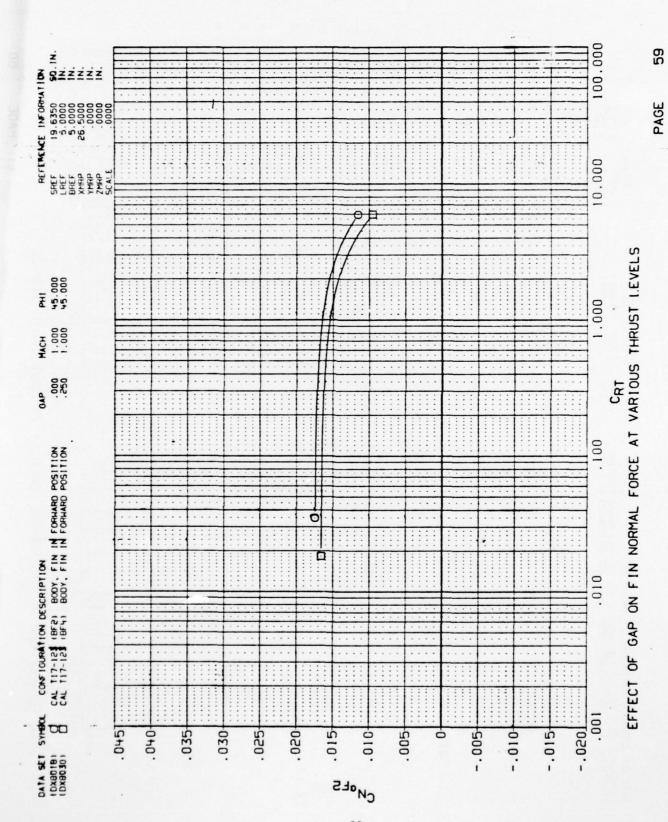


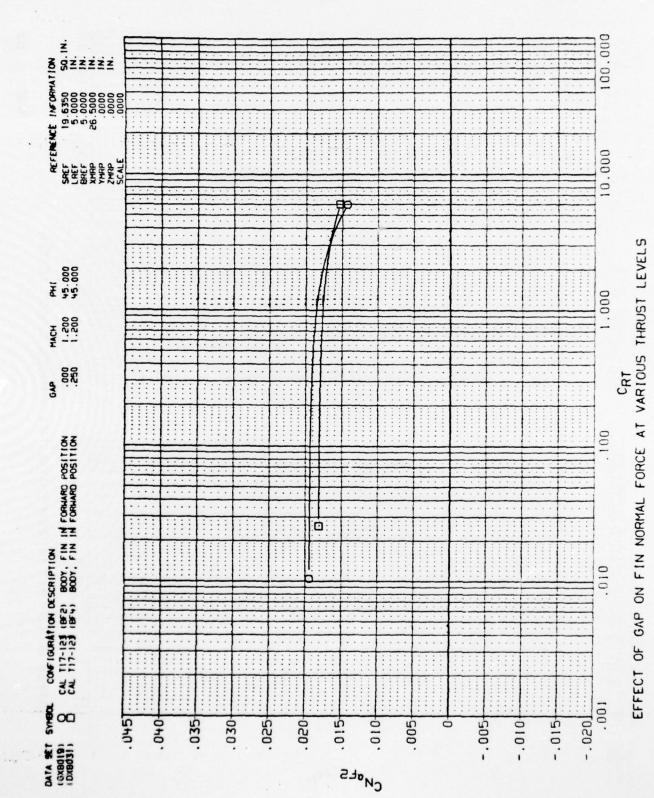












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